

Earth Observation Mission CFI Software QUICK START GUIDE

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1 ACRONYMS, NOMENCLATURE AND TERMINOLOGY

1.1 Acronyms

ANX	Ascending Node Crossing
AOCS	Attitude and Orbit Control Subsystem
ASCII	American Standard Code for Information Interchange
BOM	Beginning Of Mission
CFI	Customer Furnished Item
EOCFI	Earth Observation Customer Furnished Item
EOM	End Of Mission
ESA	European Space Agency
ESTEC	European Space Technology and Research Centre
GPL	GNU Public License
GPS	Global Positioning System
IERS	International Earth Rotation Service
I/F	Interface
LS	Leap Second
OBT	On-board Binary Time
OSF	Orbit Scenario File
SRAR	Satellite Relative Actual Reference
SUM	Software User Manual
TAI	International Atomic Time
UTC	Coordinated Universal Time
UT1	Universal Time UT1
WGS[84]	World Geodetic System 1984

1.2 Nomenclature

<i>CFI</i>	A group of CFI functions, and related software and documentation that will be distributed by ESA to the users as an independent unit
<i>CFI function</i>	A single function within a CFI that can be called by the user
<i>Library</i>	A software library containing all the CFI functions included within a CFI plus the supporting functions used by those CFI functions (transparently to the user)

1.3 Note on Terminology

In order to keep compatibility with legacy CFI libraries, the Earth Observation Mission CFI Software makes use of terms that are linked with missions already or soon in the operational phase like the Earth Explorers.

This may be reflected in the rest of the document when examples of Mission CFI Software usage are proposed or description of Mission Files is given.

2 APPLICABLE AND REFERENCE DOCUMENTS

2.1 Reference Documents

[MCD]	Earth Observation Mission CFI Software. Conventions Document. EO-MA-DMS-GS-0001. Issue 4.8 29/10/2014
[MSC]	Earth Observation Mission CFI Software. Mission Specific Customizations. EO-MA-DMS-GS-0018. Issue 4.8 29/10/2014.
[GEN_SUM]	Earth Observation Mission CFI Software. General Software User Manual. EO-MA-DMS-GS-0002. Issue 4.8 29/10/2014
[F_H_SUM]	Earth Observation Mission CFI Software. EO_FILE_HANDLING Software User Manual. EO-MA-DMS-GS-0008. Issue 4.8 29/10/2014
[D_H_SUM]	Earth Observation Mission CFI Software. EO_DATA_HANDLING Software User Manual. EO-MA-DMS-GS-0007. Issue 4.8 29/10/2014
[LIB_SUM]	Earth Observation Mission CFI Software. EO_LIB Software User Manual. EO-MA-DMS-GS-0003. Issue 4.8 29/10/2014
[ORB_SUM]	Earth Observation Mission CFI Software. EO_ORBIT Software User Manual. EO-MA-DMS-GS-0004. Issue 4.8 29/10/2014
[PNT_SUM]	Earth Observation Mission CFI Software. EO_POINTING Software User Manual. EO-MA-DMS-GS-0005. Issue 4.8 29/10/2014
[VIS_SUM]	Earth Observation Mission CFI Software. EO_VISIBILITY Software User Manual. EO-MA-DMS-GS-0006. Issue 4.8 29/10/2014

The latest applicable version of [MCD], [F_H_SUM], [D_H_SUM], [LIB_SUM], [ORB_SUM], [POINT_SUM], [VISIB_SUM], [GEN_SUM] is v4.10 and can be found at: http://eop-cfi.esa.int/REPO/PUBLIC/DOCUMENTATION/CFI/EOCFI/BRANCH_4X/

3 INTRODUCTION

3.1 Functions Overview

The Earth Observation Mission CFI Software is a collection of software functions performing accurate computations of mission related parameters for Earth Observation missions. The functions are delivered as six software libraries gathering functions that share similar functionalities:

- EO_FILE_HANDLING: functions for reading and writing files in XML format.
- EO_DATA_HANDLING: functions for reading and writing Earth Observation Mission files.
- EO_LIB: functions for time transformations, coordinate transformations and other basic transformations.
- EO_ORBIT: functions for computing orbit information.
- EO_POINTING: functions for pointing calculations.
- EO_VISIBILITY: functions for getting visibility time segments of the satellite.

A detailed description about the software can be found in the user manuals (see section 2), a general overview and information about how to get and install the software is in [GEN_SUM] while detailed function description appears in the other user manuals, one per library. It is highly recommended to read [GEN_SUM] before going ahead with the current document.

The purpose of the current document is to give complementary information to the user manuals to provide a general view of what the Earth Observation CFI Software can do and the strategies to follow for the different use cases

4 EARTH OBSERVATION CFI USAGE

The usage cases of the CFI can be classified in the following categories:

- Reading XML files
- Writing XML files
- Reading/writing Earth Observation Mission files
- Verifying XML files
- Time correlation initialisation
- Time transformations
- Other time calculations
- Using different astronomical models
- Coordinate transformations
- Orbit initialisation
- Orbital calculations
- Orbit propagation
- Orbit interpolation
- Generation of Earth Observation Mission Orbit Files
- Target calculation:
 - Attitude initialisation.
 - Atmosphere initialisation.
 - DEM
- Swath calculations
- Visibility calculations
- Time segments manipulation

In the following sections, each case is described together with the strategy to follow to get the desired results. For each case, a set of examples is provided. Besides these examples, there is a C-program example per library that is distributed with the CFI installation package (see [GEN_SUM]section 6.6)

4.1 CFI Identifiers (Ids)

Before continuing with the usage cases, it is useful to understand what are the CFI Identifiers (from now on, they will be noted as Ids).

In most cases, CFI functions need to make use of a certain amount of internal data that characterise the system. The way to provide this data to the functions is a variable, the Id. In fact the Id is just a structure that contains all the needed internal data.

Different kinds of Ids have been created to reflect the different categories or "objects" that group the data handled in the CFI. This means that each Id type stores internal data needed for a specific computation. The data stored in the Ids are hidden from the user, however the data can be accessed through a set of specific functions that retrieve the information from the Ids (see the Software User Manuals in section 2).

A list of the Ids used in the CFI is given in the table below:

Table 1: CFI identifiers

ID	Library	Description	Usage	Dependencies
sat_id		Satellite identifier	Input parameter (does not need to be initialised)	Independent, no previous initialisation of any other Id is required
x1_model_id	EO_LIB	It stores the data about the models to be used for astronomical models	Output parameter in the initialisation. Input parameter for model-dependent functions and closing function	Independent, no previous initialisation of any other Id is required. A no-initialised ID can be used
x1_time_id	EO_LIB	It stores the time correlations	Output parameter in the initialisation. Input parameter for time related computations and time closing function	Independent, no previous initialisation of any other Id is required
xo_orbit_id	EO_ORBIT	It stores the orbit data needed for orbit calculations	Output parameter in the initialisation. Input parameter in orbit calculations, propagation, interpolation, visibility computations and orbit closing function	$xo_orbit_id = sat_id + time_id + (orbit\ data)$. It requires that x1_time_id has been previously initialised
xp_atmos_id	EO_POINTING	It stores the atmospheric data used in target functions	Output parameter in the initialisation. Input parameter in target routines and atmospheric closing function	Independent, no previous initialisation of any other Id is required
xp_dem_id	EO_POINTING	It stores the Digital Elevation Model data used in target functions	Output parameter in the initialisation. Input parameter in target routines and DEM closing function	Independent, no previous initialisation of any other Id is required
xp_sat_nom_trans_id	EO_POINTING	It stores the Satellite Nominal Attitude Ref. Frame data used in attitude functions	Output parameter in the initialisation. Input parameter in attitude routines and satellite nominal attitude transformation closing function	Independent, no previous initialisation of any other Id is required, except when the file initialisation routine is used. Then xp_sat_nom_trans_id requires that x1_time_id has been previously

				initialised
xp_sat_att_trans_id	EO_POINTING	It stores the Satellite Attitude Ref. Frame data used in attitude functions	Output parameter in the initialisation. Input parameter in attitude routines and satellite attitude transformation closing function	Independent, no previous initialisation of any other Id is required, except when the file initialisation routine is used. Then xp_sat_att_trans_id requires that xl_time_id has been previously initialised
xp_instr_trans_id	EO_POINTING	It stores the Instrument Ref. Frame data used in attitude functions	Output parameter in the initialisation. Input parameter in attitude routines and instrument transformation closing function	Independent, no previous initialisation of any other Id is required, except when the file initialisation routine is used. Then xp_instr_trans_id requires that xl_time_id has been previously initialised
xp_attitude_id	EO_POINTING	It stores the results of the attitude calculation used in target functions	Output parameter in the initialisation. Input parameter in target routines and attitude closing function	$xp_attitude_id = xl_time_id + xp_sat_nom_trans_id + xp_sat_att_trans_id + xp_instr_trans_id +$ attitude computation. It requires that xl_time_id has been previously initialised but it does not necessary require that xp_sat_nom_trans_id, xp_sat_att_trans_id and xp_instr_trans_id have been previously initialised
xp_target_id	EO_POINTING	It stores the results of the target calculation, needed to get ancillary results	Output parameter in the initialisation. Input parameter in extra results target routines and target closing function	$xp_target_id = xp_attitude_id + xp_atmos_id + xp_dem_id +$ target data. It requires that xp_attitude_id has been previously initialised but, it does not necessary require that xp_dem_id and xp_atmos_id_id have been previously initialised
run_id	all	It stores a set of Ids.	It is used for calling functions with simplified interfaces as only the run_id has to be provided	Independent, but all ids that are included in the run_id depend on it, so the run_id has to be freed before the run id.

Note that the last entry in the table is an Id, called *runId*, that includes a group of Ids. All functions that has an Id in the interface, has a equivalent interface that replaces all the Ids for the *run_id*. This equivalent function has the same name that the original one but ended with the suffix *_run*.

Next figure shows the dependency between the Ids.

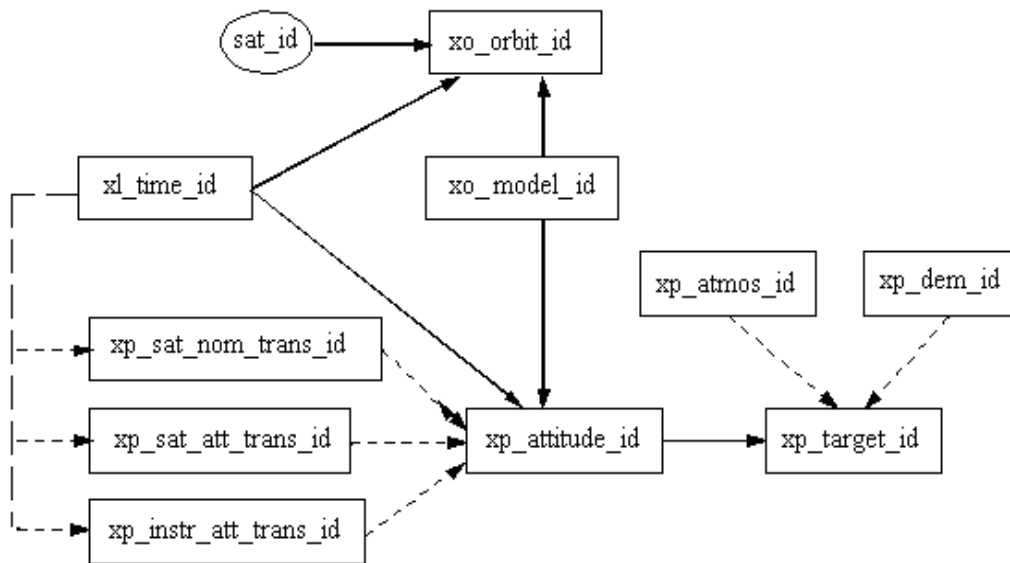


Figure 1: Hierarchical structure of the initialisation variables in the CFI

To get a complete description of the Ids, refer to [GEN_SUM].

4.2 Error Handling

A complete description of the error handling for the Earth Observation CFI functions can be found in [GEN_SUM] section 8.

4.3 Reading XML files

The CFI provides a set of functions for reading XML files, all they within the EO_FILE_HANDLING library

The strategy to read a file is the following:

- Open the file (with **xf_tree_init_parser**): note that this function returns a number that identifies the file. Every time a file is open, a new number is assigned to the file. The maximum number of XML files that can be opened is 10.
- Read values from the file: The file has to be identified with the number provided by the previous function. There are several ways of reading the file:
 - Sequentially
 - Random access
- Close the file (with **xf_tree_cleanup_parser**)

A detailed description of the reading process can be found in [F_H_SUM].

Example 4.3 - I: Reading XML files

<pre>long fd, error; char xmlFile[] = "my_xml_file"; char string_element[] = "First_Tag"; char string_value[256]; ...</pre>	Variables declaration
<pre>/* Open file */ fd = xf_tree_init_parser (xmlFile, &error); if (error < XF_CFI_OK) { printf("\nError parsing file %s\n", xmlFile); return (-1) }</pre>	Open File
<pre>/* Read the string element value in <First_Tag> */ xf_tree_read_string_element_value (&fd, string_element, string_value, &error); if (error < XF_CFI_OK) { printf("\nError reading element as string\n"); } else { printf ("Element: %s *** Value: %s\n", string_element, string_value); }</pre>	Reading routines
<pre>/* Close file */ xf_tree_cleanup_parser (&fd, &error); if (error < XF_CFI_OK) { printf("\nError freeing file %s\n", xmlFile); return(-1); }</pre>	Close File

4.4 Writing XML files

The CFI provides a set of functions for writing XML files, all they within the EO_FILE_HANDLING library.

The strategy to write a file is the following:

- Create the file (with `xf_tree_create`): note that this function returns a number that identifies the file. Every time a file is open, a new number is assigned to the file. The maximum number of XML files that can be opened simultaneously is 10.
- Write values in the file: The file has to be identified with the number provided by the previous function.
- Write file to disk (with `xf_tree_write`)
- Close the file (with `xf_tree_cleanup_parser`)

A detailed description of the reading process can be found in [F_H_SUM].

Example 4.4 - I: Writing XML files from scratch

```
/* Variables declaration */
long fd, error;
char xmlFile[] = "my_xml_file";
...
```

Variables
declaration

```
/* Create the file parser */
fd = xf_tree_create (&error);
if ( error < XF_CFI_OK )
{
    printf("\nError parsing file \n");
    return (-1);
}
```

Create file
structure

```
/* Create the root element */
xf_tree_create_root (&fd, "Earth_Explorer_File", &error);
if ( error < XF_CFI_OK )
{
    printf("\nError creating file \n");
    return (-1);
}

/* Add a child to the root element */
xf_tree_add_child (&fd, "/Earth_Explorer_File", "First_Tag", &error );
if ( error < XF_CFI_OK )
{
    printf("\nError adding adding a child \n" );
}

/* Add a value to the "First_Tag" */
xf_tree_set_string_node_value ( &fd, ".", "value_1", "%s", &error );
if ( error < XF_CFI_OK )
{
    printf("\nError adding adding a child \n" );
}
```

Writing routines

```
/* Add a child to the root element */  
xf_tree_add_next_sibling (&fd, ".", "Second_tag", &error );  
if ( error < XF_CFI_OK )  
{  
    printf("\nError adding adding a child \n" );  
}  
  
xf_tree_set_string_node_value ( &fd, ".", "value_2", "%s", &error );  
if ( error < XF_CFI_OK )  
{  
    printf("\nError adding adding a child \n" );  
}
```

Writing routines

```
/* Write the file to disk */  
xf_tree_write (&fd, xmlFile, &error );  
if ( error < XF_CFI_OK )  
{  
    printf("\nWriting Error\n" );  
    return(-1);  
}
```

Write file to disk

```
/* Close file parser */  
xf_tree_cleanup_parser (&fd, &error);  
if ( error < XF_CFI_OK )  
{  
    printf("\nError freeing file %s\n", xmlFile);  
    return(-1);  
}  
  
[...]
```

Close File

The resulting file would be as follows:

```
<?xml version="1.0"?>  
<Earth_Explorer_File>  
    <First_tag>value_1</First_tag>  
    <Second_tag>value_2</Second_tag>  
</Earth_Explorer_File>
```

4.5 Reading/Writing Earth Observation files

The Earth Observation CFI also provides functions for reading and writing the mission files. This way by calling a single function, we can get the content of a file stored in a structure (for the reading case), or we can dump the content of a data structure to a mission file (for the writing case). The following files are supported:

- IERS Bulletin B files
- Orbit files
- Orbit Scenario files
- DORIS Navigator files
- Attitude files
- Star tracker files
- Digital Elevation files (ACE model)
- Swath Definition files
- Swath Template files
- Zone Database files
- Station Database files
- Star Database files

The versions of these Earth Observation files that are currently supported for reading and writing are described in [D_H_SUM].

All these functions are provided in the EO_DATA_HANDLING library ([D_H_SUM]).

When reading files, the user should be aware that:

- Many of the structures used for reading files contain dynamic data that is allocated within the reading function. In these cases, the memory has to be freed when it is not going to be used any more by calling the suitable function.
- The reading functions for each of the file types, does not read the fixed header. The fixed header could be read independently using the CFI function **xd_read_fhr**.
- When reading the fixed header with **xd_read_fhr**, the schema name is not read (the “schema” element in the output structure **xd_fhr** will be set to “_NOSCHEMA_”). If required, the schema name and version should be read independently with the CFI functions in `explorer_file_handling`.

When writing files, the user should be aware that:

- The schema name and version can be written in the file in the following ways:
 - Setting the schema name in the “schema” element in the **xd_fhr** structure. When calling the **xd_write_xxx** function, the schema name and version will be written in the file. Note that if the schema name is set to “_NOSCHEMA_”, the schema attributes will not be written in the file.
 - After writing the file, by calling the function **xf_set_schema** (in `explorer_file_handling`).
- The CFI function **xd_select_schema** allows to get the default schema name with which the file to be written is compliant.

Example 4.5 - I: Reading and writing an Orbit Scenario file

```
long status, func_id, n;
long ierr[XD_NUM_ERR_READ_OSF];
char msg[XD_MAX_COD][XD_MAX_STR];
char input_file[] = "OSF_File.EEF"
char output_file[] = "Copy_of_OSF_File.EEF"
xd_osf_file osf_data;
```

Variables
declaration

```
/* reading OSF file */
status = xd_read_osf(input_file, &osf_data, ierr);

/* error handling */
if (status != XD_OK)
{
    func_id = XD_READ_OSF_ID;
    xd_get_msg(&func_id, ierr, &n, msg);
    xd_print_msg(&n, msg);
    if (status <= XD_ERR) return(XD_ERR);
}
```

Open File

```
/* Print results */

printf("- Number of records      : %ld ", osf_data.num_rec);
printf("- 1st. Orbital Change: \n");
printf("      Absolute Orbit: %ld\n", osf_data.osf_rec[0].abs_orb);
printf("      Cycle days      : %ld\n", osf_data.osf_rec[0].cycle_days);
printf("      Cycle orbits    : %ld\n", osf_data.osf_rec[0].cycle_orbits);
printf("      MLST           : %f\n", osf_data.osf_rec[0].mlst);

[...]
```

Using data
structure

```
/* Writing the OSF file */
status = xd_write_osf(output_file, &fhr, &osf_data, ierr);

/* error handling */
if (status != XD_OK)
{
    func_id = XD_WRITE_OSF_ID;
    xd_get_msg(&func_id, ierr, &n, msg);
    xd_print_msg(&n, msg);
    if (status <= XD_ERR) return(XD_ERR);
}
```

Writing another OSF with
the same data

```
[...]

/* Free memory */
xd_free_osf(&osf_data);
```

Free data
structure

4.6 Verifying XML files

Most of Earth Observation files are in XML format. The formats of the files are described in [D_H_SUM]. It is possible to check the format of a file with respect to its XSD schema by calling the function `xd_xml_validate` or using the standalone executable `xml_validate`. Note that

- The file can be validated using the default schema that is written in the root tag of the file.
- Or it can be validated specifying another schema in the interface of the function.

Note also that it is possible to get the last supported schema name used by the current CFI version by calling the function `xd_select_schema`.

Following there are two examples showing the use of this function. For a detailed explanation about these functions refer to [D_H_SUM].

Example 4.6 - I: Validating a file with respect to a given schema

```

/* Variables */
char input_file[256],
      schema[256],
      log_file[256];
long mode, valid_status;

strcpy (input_file, "../data/CRYOSAT_XML_OSF");
mode = XD_USER_SCHEMA;
strcpy(schema, "../..../files/schemas/EO_OPER_MPL_ORBSCT_0100.XSD");
strcpy(logfile, ""); /* => Show the validation outputs in the standard output */

```

Variable declaration & Initialisation

```

/* Validate the file */
status = xd_xml_validate (input_file, &mode, schema, logfile,
                          &valid_status, ierr);

/* error handling */
if (status != XD_OK)
{
    func_id = XD_XML_VALIDATE_ID;
    xd_get_msg(&func_id, ierr, &n, msg);
    xd_print_msg(&n, msg);
    if (status <= XD_ERR) return(XD_ERR);
}

/* Print output values */
printf("Validation status for %s: [%s]\n", input_file,
       (valid_status == XD_OK)? "VALID" : "INVALID");

```

File validation

Example 4.6 - II: Validating a file with respect to the default schema

```

strcpy(schema, "");
mode = XD_DEFAULT_SCHEMA; /* The schema is taken from the root element
                           in the file*/

/* Validate the file */
status = xd_xml_validate (input_file, &mode, schema, logfile,
                          &valid_status, ierr);

/* error handling */
if (status != XD_OK)

```

```
{  
    func_id = XD_XML_VALIDATE_ID;  
    xd_get_msg(&func_id, ierr, &n, msg);  
    xd_print_msg(&n, msg);  
    if (status <= XD_ERR) return(XD_ERR);  
}  
  
/* Print output values */  
printf("Validation status for %s: [%s]\n", input_file,  
       (valid_status == XD_OK)? "VALID" : "INVALID");
```

4.7 Configure position/attitude interpolator (decimation)

The list of orbit state vectors and attitude records can be configured according to user need. This can be done in structures corresponding to orbit or attitude files, using the corresponding functions (see [D_H_SUM] for detailed explanation):

- `xd_orbit_file_decimate`, for orbit files.
- `xd_attitude_file_decimate`, for attitude files.

These functions decimate the input record list according to input decimate-delta time.

Example 4.7: Configuring orbit file interpolator

```
/* Variables */
xd_fhr fhr_in, fhr_out;
xd_orbit_file osv_in, osv_out;
double decimation_delta_time;
long status;

[Here read orbit file (osv_in) and fixed header (fhr_in) as
explained in section 4.5]
```

Variable declaration &
Initialisation

```
/* Decimate the file */
decimation_delta_time = 5.; /* seconds */
status = xd_orbit_file_decimate(&fhr_in, &osv_in, decimation_delta_time,
                                &fhr_out, &osv_out, ierr);

/* error handling */
if (status != XD_OK)
{
    func_id = XD_ORBIT_FILE_DECIMATE_ID;
    xd_get_msg(&func_id, ierr, &n, msg);
    xd_print_msg(&n, msg);
    if (status <= XD_ERR) return(XD_ERR);
}
```

File validation

4.8 Time correlation initialisation

The initialisation of the time correlations does not provide any direct functionality to the user, but it is needed for many other operations within the mission planning.

The initialisation consist on storing the time correlation between the different allowed time references, (i.e. TAI, UTC, UT1 and GPS time) in a *xl_time_id* structure.

In order to accomplish such correlations, two possible strategies can be used:

- Initialization from a single or multiple orbit files (**xl_time_ref_init_file**).
- Initialization from a structure data containing data read from files or user data (**xl_time_id_init**).
- Initialization from a given set of time references (**xl_time_ref_init**).

After finalising the transformations, the *xl_time_id* must be freed (**xl_time_close**).

Next figure represents the data flow for the *xl_time_id* structure.

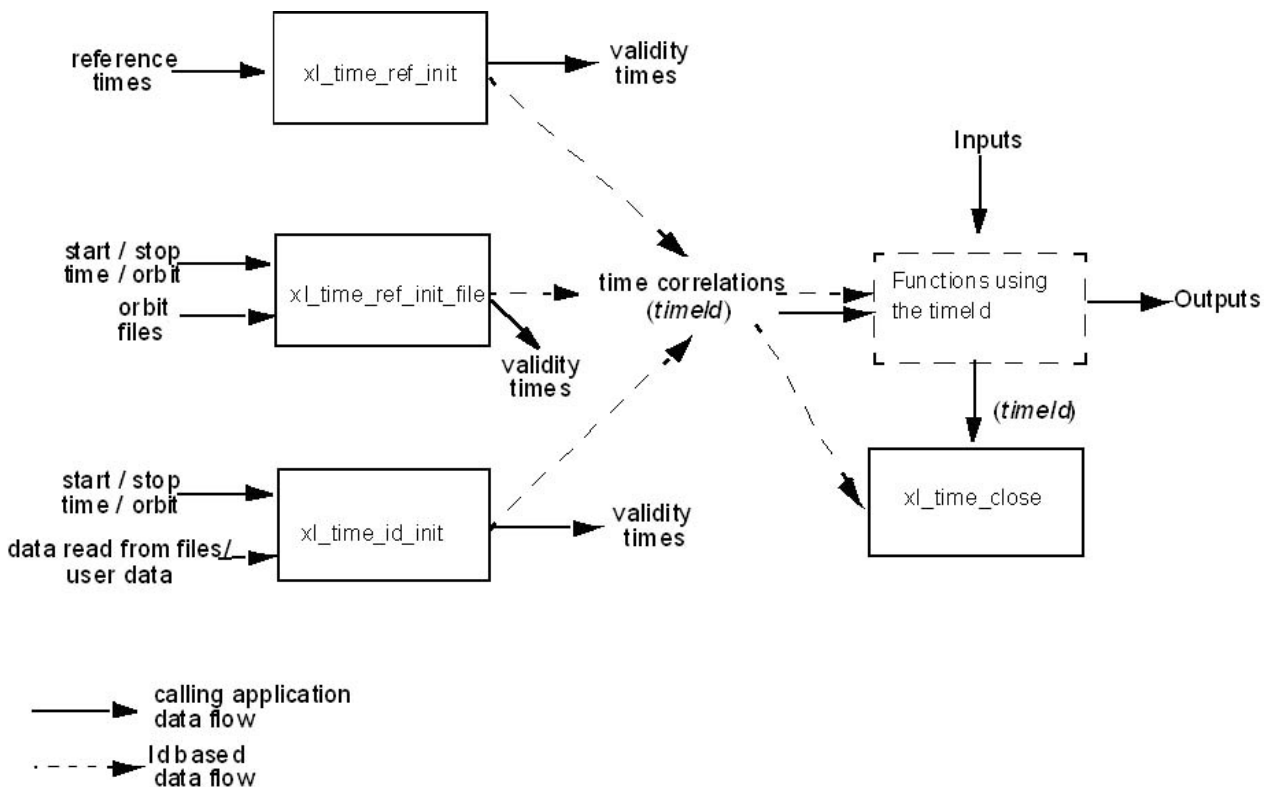


Figure 2: *xl_time_id* data flow

Examples showing the usage of the time initialization can be found in section 4.9.

4.9 Time transformations

The Earth Observation CFI Software contains a set of functions to transform an input time in a given time reference and format to another time reference and/or format.

Time transformations functions requires the user to initialise the time correlations if the time reference is going to be changed¹(see section 4.8). Once the initialisation has been performed, the user is able to transform any date expressed in one of the allowed time references to another, through the Time Format / Reference Transformation functions. The *xl_time_id* has to be provided to each of these functions. The process can be repeated as needed without initialising the time correlations each time.

For a complete description of all the time transformation function refer to [LIB_SUM].

Besides the time transformation functions, there exists a program called **time_conv** that performs the same calculation (see Example 4.8 - III)

Example 4.8 - I: Time transformations. Initialization with an IERS file

```

/* Variables */
long   status, func_id, n;
long   xl_ierr[XL_ERR_VECTOR_MAX_LENGTH];
char   msg[XL_MAX_COD][XL_MAX_STR];

xl_time_id time_id = {NULL};

long   time_model, n_files, time_init_mode, time_ref;
char   *time_file[2];
double time0, time1, val_time0, val_time1;
long   orbit0, orbit1;
long   ierr[XL_NUM_ERR_TIME_REF_INIT_FILE];
char   iers_file[] = "../data/bulb.dat";

long   format_in,   ref_in,
       format_out, ref_out;
long   transport_in[4];
char   ascii_in[XD_MAX_STR], ascii_out[XD_MAX_STR];
double proc_out;
    
```

Variable declaration

```

/* Time initialisation */
time_model   = XL_TIMEMOD_IERS_B_PREDICTED;
n_files      = 1;
time_init_mode = XL_SEL_TIME;
time_ref     = XL_TIME_TAI;
time0        = 240.0;
time1        = 260.0;
orbit0       = 0; /* dummy */
orbit1       = 0; /* dummy */
time_file[0] = iers_file;

status = xl_time_ref_init_file (&time_model, &n_files, time_file,
                               &time_init_mode, &time_ref, &time0, &time1,
                               &orbit0, &orbit1, &val_time0, &val_time1,
                               &time_id, xl_ierr);

/* error handling */
if (status != XL_OK)
    
```

Time Initialisation

1 When the output time reference is equal to the input one, there is no need of initialiasing the *time_id*

```
{
  func_id = XL_TIME_REF_INIT_FILE_ID;
  xl_get_msg(&func_id, xl_ierr, &n, msg);
  xl_print_msg(&n, msg);
  if (status <= XL_ERR) return(XL_ERR);
}
```

```
/* 1st. Time transformation: time in TAI and standard transport format to
   GPS time in standard ASCII format */
format_in = XL_TRANS_STD;
ref_in = XL_TIME_TAI;
format_out = XL_ASCII_STD_REF_MICROSEC;
ref_out = XL_TIME_GPS;

transport_in[0] = 245; /* TAI time [integer days] */
transport_in[1] = 150; /* TAI time [integer seconds] */
transport_in[2] = 1500; /* TAI time [integer microseconds] */
transport_in[3] = 0; /* Unused in Transport_Standard */

status = xl_time_transport_to_ascii(&time_id,
                                   &format_in, &ref_in, transport_in,
                                   &format_out, &ref_out, ascii_out,
                                   xl_ierr);

/* error handling */
if (status != XL_OK)
{
  func_id = XL_TIME_TRANSPORT_TO_ASCII_ID;
  xl_get_msg(&func_id, t2a_ierr, &n, msg);
  xl_print_msg(&n, msg);
  if (status <= XL_ERR) return(XL_ERR);
}

/* Print input/output values */
printf("- Transport input format: %ld \n", format_in);
printf("- Input time_reference : %ld \n", ref_in);
printf("- Input transport time : %ld, %ld, %ld \n",
       transport_in[0], transport_in[1], transport_in[2]);
printf("- ASCII input format : %ld \n", format_out);
printf("- Output time reference : %ld \n", ref_out);
printf("- Output ASCII time : %s \n", ascii_out);

/* 2nd. Time transformation: time in GPS and standard ASCII format to
   processing format and UT1 time reference */
format_in = format_out;
ref_in = ref_out;
format_out = XL_PROC;
ref_out = XL_TIME_UT1;
strcpy(ascii_in, ascii_out);

status = xl_time_ascii_to_processing(&time_id,
                                   &format_in, &ref_in, ascii_in,
                                   &format_out, &ref_out, proc_out,
                                   xl_ierr);

/* error handling */
if (status != XL_OK)
{
  func_id = XL_TIME_ASCII_TO_PROCESSING_ID;
  xl_get_msg(&func_id, t2a_ierr, &n, msg);
}
```

```

xl_print_msg(&n, msg);
if (status <= XL_ERR) return(XL_ERR);
}
    
```

[...]

```

/* Close time references */
status = xl_time_close(&time_id, xl_ierr);
if (status != XL_OK)
{
    func_id = XL_TIME_CLOSE_ID;
    xl_get_msg(&func_id, xl_ierr, &n, msg);
    xl_print_msg(&n, msg);
    if (status <= XL_ERR) return(XL_ERR);
}
    
```

Close Time
correlation

Example 4.8 - II: Time transformations. Initialisation with given time correlations.

```

/* Variables */
long status, func_id, n;
long xl_ierr[XL_ERR_VECTOR_MAX_LENGTH];
char msg[XL_MAX_COD][XL_MAX_STR];
double tri_time[4];
double tri_orbit_num, tri_anx_time, tri_orbit_duration;
xl_time_id time_id = {NULL};

long format_in, format_out,
    ref_in, ref_out;
double proc_in;
    
```

Variable declaration

```

/* Time initialisation */
tri_time[0] = -245.100000000; /* TAI time [days] */
tri_time[1] = tri_time[0] - 35.0/86400.; /* UTC time [days] (= TAI - 35.0 s) */
tri_time[2] = tri_time[0] - 35.3/86400.; /* UT1 time [days] (= TAI - 35.3 s) */
tri_time[3] = tri_time[0] - 19.0/86400.; /* GPS time [days] (= TAI - 19.0 s) */

tri_orbit_num = 10;
tri_anx_time = 5245.123456;
tri_orbit_duration = 6035.928144;

status = xl_time_ref_init(tri_time, &tri_orbit_num, &tri_anx_time,
    &tri_orbit_duration, &time_id, xl_ierr);

/* error handling */
if (status != XL_OK)
{
    func_id = XL_TIME_REF_INIT_ID;
    xl_get_msg(&func_id, xl_ierr, &n, msg);
    xl_print_msg(&n, msg);
    if (status <= XL_ERR) return(XL_ERR);
}
    
```

Time Initialisation

```

/* time from TAI to UT1 time reference in processing format */
format_in = XL_PROC;
ref_in = XL_TIME_TAI;
format_out = XL_PROC;
ref_out = XL_TIME_UT1;
proc_in = 0.0;
    
```

Time Operations

```
status = xl_time_processing_to_processing(&time_id,  
                                         &format_in, &ref_in, proc_in,  
                                         &format_out, &ref_out, proc_out,  
                                         xl_ierr);  
  
/* error handling */  
if (status != XL_OK)  
{  
    func_id = XL_TIME_PROCESSING_TO_PROCESSING_ID;  
    xl_get_msg(&func_id, t2a_ierr, &n, msg);  
    xl_print_msg(&n, msg);  
    if (status <= XL_ERR) return(XL_ERR);  
}
```

Time Operations

[...]

```
/* Close time references */  
status = xl_time_close(&time_id, xl_ierr);  
if (status != XL_OK)  
{  
    func_id = XL_TIME_CLOSE_ID;  
    xl_get_msg(&func_id, xl_ierr, &n, msg);  
    xl_print_msg(&n, msg);  
    if (status <= XL_ERR) return(XL_ERR);  
}
```

Close Time
correlation

Example 4.8 - III: Time transformation with executable file.

The following command line does the same transformation than the code in :

```
time_conv -fmt_in PROC -fmt_out PROC -ref_in TAI -ref_out UT1 -day 0.0 -v  
          -tai 0.0000 -gps 0.00021991 -utc 0.00040509 -ut1 0.00040865
```

4.10 Other time calculations

Besides the time transformation functions shown in section , the CFI provide functions for:

- Operation between Dates
 - **xl_time_add**: adds a duration to a TAI, UTC, UT1 or GPS time expressed in Processing format.
 - **xl_time_diff**: subtracts two TAI, UTC, UT1 or GPS times expressed in Processing format.
- Transformations from/to On-board Times
 - **xl_time_obt_to_time**: transforms an On-board Time (OBT) into a TAI, UTC, UT1 or GPS time in processing format.
 - **xl_time_time_to_obt**: transforms a TAI, UTC, UT1 or GPS time expressed in Processing format into an On-board Time (OBT).

These functions do not need to follow any special strategy and can be called from any part of the program without having to initialise the timeId.

Example 4.9 - I: Adding two dates

```

/* Variables */
long   status, func_id, n;
long   xl_ierr[XL_ERR_VECTOR_MAX_LENGTH];
char   msg[XL_MAX_COD][XL_MAX_STR];

double proc_1, proc_2, proc_out;
long   proc_id, time_ref;

proc_id = XL_PROC;
time_ref = XL_TIME_TAI;
proc_1 = 245.100001; /* Processing Time, MJD2000 [days] */
proc_2 = 110.123456; /* Added duration [days] */

/* Call xl_time_add function */
status = xl_time_add(&proc_id, &time_ref, &proc_1, &proc_2,
                    &proc_out, xl_ierr);

/* Error handling */
if (status != XL_OK)
{
    func_id = XL_TIME_ADD_ID;
    xl_get_msg(&func_id, tad_ierr, &n, msg);
    xl_print_msg(&n, msg);
    if (status <= XL_ERR) return(XL_ERR);
}

/* Print output values */
printf("- Output time (TAI) = %12.12lf + %12.12lf = %12.12lf days",
        proc_1, proc_2, proc_out);
[...]
```

4.11 Using different astronomical models

The EOCFI software allows the user to choose the models for the Earth shape and astronomical calculations. The models that can be chosen are grouped in the following categories (for further details refer to [LIB_SUM]):

- Earth
- Sun
- Moon
- Planet
- Star
- Nutation
- Precession
- Physical and astronomical constants

In order to select the models with which the EOCFI has to work, a CFI ID called `model_id` has been created (see [GEN_SUM], section 7.3). The calling sequence for a C program where the `model_id` is needed, would be as follows:

- Declare the `model_id` variable:
xl_model_id `model_id = {NULL};`
- The `model_id` has to be initialised this way (as other CFI ID's), so that the EOCFI could recognise that the `model_id` is not initialised.
- Optionally, initialise the `model_id` with **xl_model_init**. This function would set the requested models in the `model_id`. If the `model_id` is not initialised, the EOCFI functions will use the default models.
- The `model_id` is used as an input parameter in the EOCFI functions if it is needed.
- Close the `model_id` with **xl_model_close** (Only if the `model_id` was initialised).

This strategy can be seen in the Example 4.11-I. For other examples, the default models will be used (`model_id` no initialised)

4.12 Coordinate transformations

The Earth Observation CFI software provides a set of functionality for coordinate transformations:

- Transformations between reference frames: It is possible to transform between the following reference frames: Galactic, Heliocentric, Barycentric Mean of 1950, Barycentric Mean of 2000, Geocentric Mean of 2000, Mean of Date, True of Date, Earth Fixed, Topocentric.
- These transformations are carried out by the following functions: **xl_change_cart_cs**, **xl_topocentric_to_ef** and **xl_ef_to_topocentric**.
- Transformations between Euler's angles and its equivalent rotation matrix (**xl_euler_to_matrix** and **xl_matrix_to_euler**)
- Rotate vectors and compute the rotation angles between two orthonormal frames (**xl_get_rotated_vectors** and **xl_get_rotation_angles**).
- Transformations between vectors and quaternions (**xl_quaternions_to_vectors** and **xl_vectors_to_quaternions**)
- Coordinate Transformations between Geodetic and Cartesian coordinates (**xl_geod_to_cart** and **xl_cart_to_geod**)
- Transformations between cartesian coordinates right ascension and declination angles (**xl_cart_to_radec** and **xl_radec_to_cart**)
- Transformations between Keplerian elements and Cartesian coordinates (**xl_kepl_to_cart** and **xl_cart_to_kepl**)
- Calculation of the osculating true latitude for a cartesian state vector (**xl_position_on_orbit**)

All the functions are described in [LIB_SUM].

xl_change_cart_cs and **xl_position_on_orbit**, require the time initialisation before they are called, so the strategy to follow is the same as for the time transformations functions (see section to know more about how to initialise the time correlations). The other functions do not need any special action before calling them.

Example 4.11 - I: Coordinate transformation

```

/* Variables */
long   status, func_id, n;
long   xl_ierr[XL_ERR_VECTOR_MAX_LENGTH];
char   msg[XL_MAX_COD][XL_MAX_STR];

xl_time_id  time_id = {NULL};
xl_model_id model_id = {NULL};

long   model_mode,
       models[XL_NUM_MODEL_TYPES_ENUM];
long   cs_in, cs_out;
long   calc_mode = XL_CALC_POS_VEL_ACC;
long   time_ref = XL_TIME_TAI;
double time=2456.0;

double pos[3] = {-6313910.323647, 3388282.485785, 0.002000};
double vel[3] = {531.059763, 971.331224, 7377.224410};
double acc[3] = {-0.175235, 0.095468, 0.000000};
    
```

Variable declaration

[... Time initialisation...]

```

/* Model initialisation:
   models set to default models.
   "models" array does not need to be initialised */
model_mode = XL_MODEL_DEFAULT;
status = xl_model_init(&model_mode, models,
                      &model_id, xl_ierr);
if (status != XL_OK)
{
  func_id = XL_MODEL_INIT_ID;
  xl_get_msg(&func_id, xl_ierr, &n, msg);
  xl_print_msg(&n, msg);
  if (status <= XL_ERR) return(XL_ERR);
}

```

Model
Initialisation

```

cs_in = XL_TOD; /* Initial coordinate system = True of Date */
cs_out = XL_EF; /* Final coordinate system = Earth fixed */
ext_status = xl_change_cart_cs(&model_id, &time_id,
                              &calc_mode, &cs_in, &cs_out,
                              &time_ref, &time_t, pos, vel, acc,
                              &calc_mode, &cs_in, &cs_out,
                              pos_out, vel_out, acc_out);

if (ext_status != XL_OK)
{
  func_id = XL_CHANGE_CART_CS_ID;
  xl_get_msg(&func_id, &ext_status, &n, msg);
  xl_print_msg(&n, msg);
  if (ext_status <= XL_ERR) return(XL_ERR);
}
/* Print output values */
printf("EF Position      : %lf, %lf, %lf\n",
       pos_out[0], pos_out[1], pos_out[2]);
printf("EF Velocity     : %lf, %lf, %lf\n",
       vel_out[0], vel_out[1], vel_out[2]);
printf("EF Acceleration: %lf, %lf, %lf\n",
       acc_out[0], acc_out[1], acc_out[2]);
[...]
```

Change Coordinate System

```

/* Transform to geodetic coordinates */
ext_status = xl_cart_to_geod(&model_id,
                            &calc_mode, pos_out, vel_out,
                            &lon, &lat, &h, &lond, &latd, &hd);

if (ext_status != XL_OK)
{
  func_id = XL_CART_TO_GEOD_ID;
  xl_get_msg(&func_id, &ext_status, &n, msg);
  xl_print_msg(&n, msg);
  if (ext_status <= XL_ERR) return(XL_ERR);
}

/* Print output values */
printf("- Geocentric longitude [deg]      : %lf ", lon_t);
printf("- Geodetic latitude [deg]         : %lf ", lat_t);
printf("- Geodetic altitude [m]            : %lf ", h_t);
printf("- Geocentric longitude rate [deg/s]  : %lf ", lond_t);
printf("- Geodetic latitude rate [deg/s]     : %lf ", latd_t);
printf("- Geodetic altitude rate [m/s]      : %lf ", hd_t);

```

Get geodetics coordinates


```
/* Close model initialisation */  
status = xl_model_close(&model_id, xl_ierr);  
if (status != XL_OK)  
{  
    func_id = XL_MODEL_CLOSE_ID;  
    xl_get_msg(&func_id, xl_ierr, &n, msg);  
    xl_print_msg(&n, msg);  
    if (status <= XL_ERR) return(XL_ERR);  
}
```

Close Model ID

[... Close Time initialisation...]

4.13 Orbit initialization

In order to get orbit related information it is needed to provide some data about the orbit. These data have to be stored in the **xo_orbit_id** (see section 4.1) before any other calculation involving orbital data could be done. These calculations where the **xo_orbit_id** structure are needed are:

- Transformations between time and orbit number
- Getting orbit information
- Orbit propagation and interpolation

The strategy to follow for initializing the orbit and the afterward usage can be summarize in the following steps:

- Time correlation initialization (see section 4.8): the *xl_time_id* is needed for the orbital initialisation in the next step.
- Orbital initialization (getting the *xo_orbit_id*): In this step, the user provides orbital information that will be used in further calculations. The data are stored in the *xo_orbit_id* “object”. There are three ways of initialising the orbit:
 - Providing information about the orbital geometry with **xo_orbit_init_def**.
 - Providing a osculating state vector for a given time and orbit number (see function **xo_orbit_cart_init[_precise]**).
 - Providing orbit files through the function **xo_orbit_init_file[_precise]**: The orbital files usually contain time correlation data. To ensure that orbit routines produce correct results, these same time correlations should be in the orbit file and the *xl_time_id*.
 - Providig data structures containg user data or data read from files through the function **xo_orbit_id_init**. This function is completely equivalent to **xo_orbit_init_file**.
- Orbit computations: getting orbital information, propagation, interpolation.
- Close orbital initialisation by calling **xo_orbit_close**.
- Close Time initialisation.

A whole description of the functions can be found in [ORB_SUM].

Next figure shows the data flow for the orbital calculations.

The sections 4.14 and 4.15 contain examples showing the orbit initialization usage.

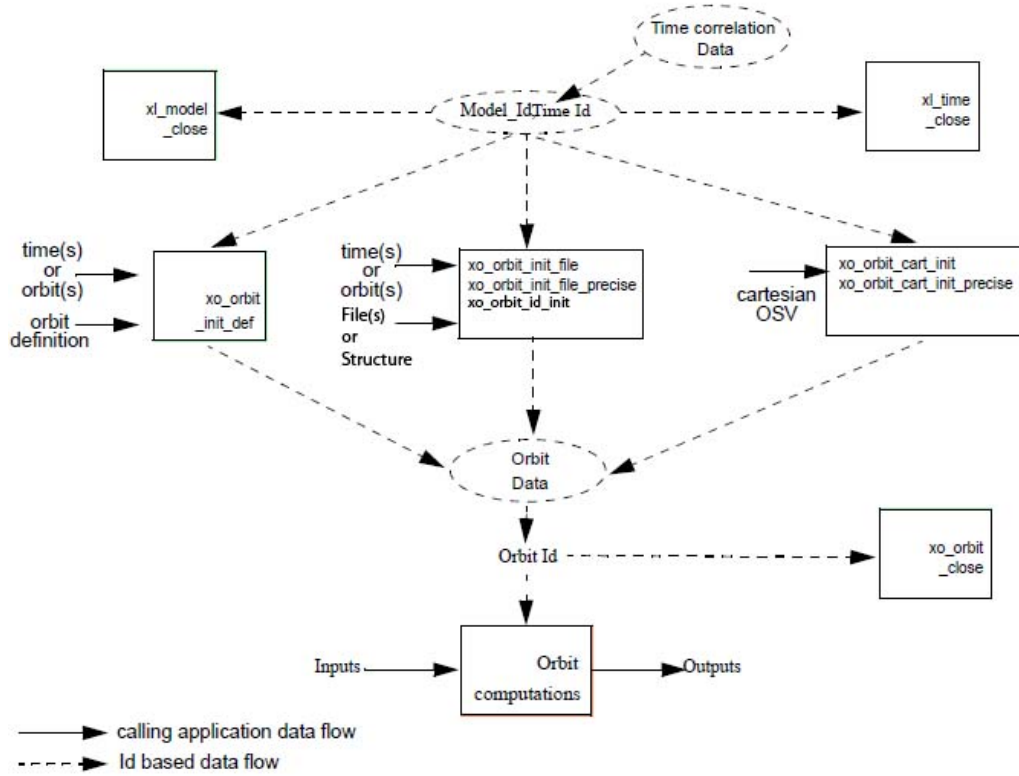


Figure 3: Orbit Information Routines Data Flow

4.14 Orbit calculations

The Earth Observation CFI functions allow to get the following orbital information for a satellite:

- Transformation between time and orbits: It is possible to know the orbit number and the time after the ANX for a given input time and viceversa (functions `xo_time_to_orbit` and `xo_orbit_to_time`)
- Orbital parameters and orbital numbers (functions `xo_orbit_info`, `xo_orbit_rel_from_abs`, `xo_orbit_abs_from_rel`, `xo_orbit_abs_from_phase`)
- Times for which an input set of Sun zenith angles are reached, Sun occultations by the Earth and Sun occultations by the Moon (function `xv_orbit_extra`). See Example 4.13 -II.

A whole description of the functions can be found in [ORB_SUM] and [VIS_SUM].

All this functions require the orbit initialisation (section 4.13). The `xo_orbit_id` can be computed with whatever initialisation function, except for the functions that compute the orbit numbers (`xo_orbit_rel_from_abs`, `xo_orbit_abs_from_rel`, `xo_orbit_abs_from_phase`), for which the `xo_orbit_id` has to be initialised with `xo_orbit_init_file` using an Orbit Scenario file.

Example 4.13 - I: Orbital calculations with `xo_orbit_init_def`

```

/* Variables */
long   status, func_id, n;
char   msg[XL_MAX_COD][XL_MAX_STR];
long   xl_ierr[XL_ERR_VECTOR_MAX_LENGTH];
long   xo_ierr[XO_ERR_VECTOR_MAX_LENGTH];

long   sat_id      = XO_SAT_CRYOSAT;
xl_time_id time_id = {NULL};
xl_model_id model_id = {NULL};
xo_orbit_id orbit_id = {NULL};

double tri_time[4];
double tri_orbit_num, tri_anx_time, tri_orbit_duration;

long   irep, icyc, iorb0, iorb;
double ascmlst, rlong, ascmlst_drift, inclination;
double time0, time;

long   abs_orbit, rel_orbit, cycle, phase;
double result_vector[XO_ORBIT_INFO_EXTRA_NUM_ELEMENTS];

long   orbit_t, second_t, microsec_t;
long   time_ref = XL_TIME.UTC;
double time_t;
    
```

Variable declaration

```

/* Time initialisation */
tri_time[0] = -245.100000000;          /* TAI time [days] */
tri_time[1] = tri_time[0] - 35.0/86400.; /* UTC time [days] (= TAI - 35.0 s) */
tri_time[2] = tri_time[0] - 35.3/86400.; /* UT1 time [days] (= TAI - 35.3 s) */
tri_time[3] = tri_time[0] - 19.0/86400.; /* GPS time [days] (= TAI - 19.0 s) */

tri_orbit_num = 10;
tri_anx_time = 5245.123456;
tri_orbit_duration = 6035.928144;

status = xl_time_ref_init(tri_time, &tri_orbit_num, &tri_anx_time,
                          &tri_orbit_duration, &time_id, xl_ierr);

/* error handling */
if (status != XL_OK)
{
  func_id = XL_TIME_REF_INIT_ID;
  xl_get_msg(&func_id, xl_ierr, &n, msg);
  xl_print_msg(&n, msg);
  if (status <= XL_ERR) return(XL_ERR); /* CAREFUL: normal status */
}

```

Time Initialisation

```

/* Orbit initialisation: xo_orbit_init_def */
irep = 369;          /* Repeat cycle of the reference orbit [days] */
icyc = 5344;        /* Cycle length of the reference orbit [orbits] */
ascmlst = 8.6667;   /* Mean local solar time at ANX [hours] */
rlong = -36.2788;  /* Geocentric longitude of the ANX [deg] */
iorb0 = 0;         /* Absolute orbit number of the reference orbit */
ascmlst_drift = -179.208556;
inclination = 0.0;

time_init_mode = XO_SEL_ORBIT;
drift_mode = XO_NOSUNSYNC_DRIFT;
time0 = -2456.0; /* UTC time in MJD2000 (1993-04-11 00:00:00) [days] */
time = 0.0; /* Dummy */

/* Calling to xo_orbit_init_def */
status = xo_orbit_init_def(&sat_id, &model_id, &time_id,
                           &time_ref, &time0, &iorb0,
                           &drift_mode, &ascmlst_drift, &inclination,
                           &irep, &icyc, &rlong, &ascmlst,
                           &val_time0, &val_time1, &orbit_id, xo_ierr);

/* error handling */
if (status != XO_OK)
{
  func_id = XO_ORBIT_INIT_DEF_ID;
  xo_get_msg(&func_id, xo_ierr, &n, msg);
  xo_print_msg(&n, msg);
}

```

Orbit Initialisation

```

/* Get orbit info */
abs_orbit = 100;
status = xo_orbit_info (&orbit_id, &abs_orbit, result_vector, xo_ierr);

/* error handlingg */
if (status != XO_OK)
{
    func_id = XO_ORBIT_INFO_ID;
    xo_get_msg(&func_id, xo_ierr, &n, msg);
    xo_print_msg(&n, msg);
}

/* print results */
printf("\n\t- Absolute orbit = %ld", abs_orbit);
printf("\n\t- Repeat cycle = %lf", result_vector[0]);
printf("\n\t- Cycle length = %lf", result_vector[1]);
[...]
```

```

/* Get time for a given Orbit and ANX time */
orbit_t = 1034;
second_t = 3000;
microsec_t = 50;

status = xo_orbit_to_time(&orbit_id,
                        &orbit_t, &second_t, &microsec_t,
                        &time_ref, &time_t, xo_ierr);

/* error handlingg */
if (status != XO_OK)
{
    func_id = XO_ORBIT_TO_TIME_ID;
    xo_get_msg(&func_id, xo_ierr, &n, msg);
    xo_print_msg(&n, msg);
}

/* Get the Orbit and ANX time from the input time*/
status=xo_time_to_orbit(&orbit_id, &time_ref, &time_t,
                      &orbit_t, &second_t, &microsec_t, xo_ierr);

/* error handlingg */
if (status != XO_OK)
{
    func_id = XO_TIME_TO_ORBIT_ID;
    xo_get_msg(&func_id, xo_ierr, &n, msg);
    xo_print_msg(&n, msg);
}

```

Orbit functions

```

/* Close orbit initialisation */
status = xo_orbit_close(&orbit_id, xo_ierr);
if (status != XO_OK)
{
    func_id = XO_ORBIT_CLOSE_ID;
    xo_get_msg(&func_id, xo_ierr, &n, msg);
    xo_print_msg(&n, msg);
}

```

Orbit close

```

/* Close time references */
status = xl_time_close(&time_id, xl_ierr);
if (status != XL_OK)
{
    func_id = XL_TIME_CLOSE_ID;
    xl_get_msg(&func_id, xl_ierr, &n, msg);
    xl_print_msg(&n, msg);
    if (status <= XL_ERR) return(XL_ERR);
}
[...]
```

Time Close

Example 4.13 - II: Orbital calculations with xo_orbit_init_file

```

/* Variables */
long    status, func_id, n;
char    msg[XL_MAX_COD][XL_MAX_STR];
long    xl_ierr[XL_ERR_VECTOR_MAX_LENGTH];
long    xo_ierr[XO_ERR_VECTOR_MAX_LENGTH];

long    sat_id      = XO_SAT_CRYOSAT;
xl_time_id time_id  = {NULL};
xl_model_id model_id = {NULL};
xo_orbit_id orbit_id = {NULL};

double tri_time[4];
double tri_orbit_num, tri_anx_time, tri_orbit_duration;

long n_files, time_mode, orbit_mode, time_ref;
char orbit_scenario_file[XD_MAX_STR];
char *files[2];

long abs_orbit, rel_orbit, cycle, phase;
double result_vector[XO_ORBIT_INFO_EXTRA_NUM_ELEMENTS];
long num_sza;
double sza, sza_up, sza_down,
        eclipse_entry, eclipse_exit,
        sun_moon_entry, sun_moon_exit;
```

Variable declaration

```

/* Time initialisation */
tri_time[0] = -245.100000000; /* TAI time [days] */
tri_time[1] = tri_time[0] - 35.0/86400.; /* UTC time [days] (= TAI - 35.0 s) */
tri_time[2] = tri_time[0] - 35.3/86400.; /* UT1 time [days] (= TAI - 35.3 s) */
tri_time[3] = tri_time[0] - 19.0/86400.; /* GPS time [days] (= TAI - 19.0 s) */

tri_orbit_num = 10;
tri_anx_time  = 5245.123456;
tri_orbit_duration = 6035.928144;

status = xl_time_ref_init(tri_time, &tri_orbit_num, &tri_anx_time,
                          &tri_orbit_duration, &time_id, xl_ierr);

/* error handling */
if (status != XL_OK)
{
    func_id = XL_TIME_REF_INIT_ID;
    xl_get_msg(&func_id, xl_ierr, &n, msg);
    xl_print_msg(&n, msg);
    if (status <= XL_ERR) return(XL_ERR);
}
}
```

Time Initialisation

```

/* Orbit initialisation: xo_orbit_init_file */
n_files = 1;
time_mode = XO_SEL_FILE;
orbit_mode = XO_ORBIT_INIT_OSF_MODE;
time_ref = XO_TIME_UT1;
strcpy(orbit_scenario_file, "../data/CRYOSAT_XML_OSF");
files[0] = orbit_scenario_file;

status = xo_orbit_init_file(&sat_id, &model_id, &time_id,
                           &orbit_mode, &n_files, files,
                           &time_mode, &time_ref,
                           &time0, &time1, &orbit0, &orbit1,
                           &val_time0, &val_time1,
                           &orbit_id, xo_ierr);

/* error handling */
if (status != XO_OK)
{
    func_id = XO_ORBIT_INIT_FILE_ID;
    xo_get_msg(&func_id, xo_ierr, &n, msg);
    xo_print_msg(&n, msg);
}

```

Orbit Initialisation

```

/* Get orbit info */
abs_orbit = 100;
status = xo_orbit_info (&orbit_id, &abs_orbit, result_vector, xo_ierr);
if (status != XO_OK)
{
    func_id = XO_ORBIT_INFO_ID;
    xo_get_msg(&func_id, xo_ierr, &n, msg);
    xo_print_msg(&n, msg);
}

/* Get orbit extra info: Note that this function uses
   as input the result_vector from xo_orbit_info */
num_sza = 2;
sza[0] = 90;
sza[1] = 80;
status = xo_orbit_extra (&orbit_id, &abs_orbit, result_vector,
                        &num_sza, sza, &sza_up, &sza_down,
                        &eclipse_entry, &eclipse_exit,
                        &sun_moon_entry, &sun_moon_exit,
                        xv_ierr);

if (status != XO_OK)
{
    func_id = XV_ORBIT_EXTRA_ID;
    xv_get_msg(&func_id, xv_ierr, &n, msg);
    xv_print_msg(&n, msg);
}

/* Get relative orbit number and phase */
status = xo_orbit_rel_from_abs (&orbit_id, &abs_orbit,
                                &rel_orbit, &cycle, &phase, xo_ierr);

/* error handling */
if (status != XO_OK)
{
    func_id = XO_ORBIT_REL_FROM_ABS_ID;
    xo_get_msg(&func_id, xo_ierr, &n, msg);
    xo_print_msg(&n, msg);
}

```

Orbit functions


```
/* Close orbit_id*/  
status = xo_orbit_close(&orbit_id, xo_ierr);  
[...]
```

Orbit
close

```
/* Close time references */  
status = xl_time_close(&time_id, xl_ierr);  
[...]
```

Time
close

4.15 State vector computation (Propagation/Interpolation)

The object of this functionality is the accurate prediction of osculating Cartesian state vectors for user requested times. It is also possible to get ancillary results such as mean and osculating Keplerian orbit state vectors, satellite osculating true latitude, latitude rate and latitude rate-rate, Sun zenith angle and many more.

The propagation/interpol strategy is the following:

- Initialise the time correlations (section 4.8)
- Orbit initialisation (section 4.13) with any of the initialization routines for orbit: **xo_orbit_init_def**, **xo_orbit_init_file** or **xo_orbit_cart_init**.
- Compute the orbital state vector for the required time by calling the function **xo_osv_compute**. The input time has to be within the validity times for the computations. These validity times can be get with the function **xo_orbit_get_osv_compute_validity**.
- Optionally, to obtain ancillary results the user might call the **xo_osv_compute_extra** function.

The following figure shows the data flow for the computation of state vectors:

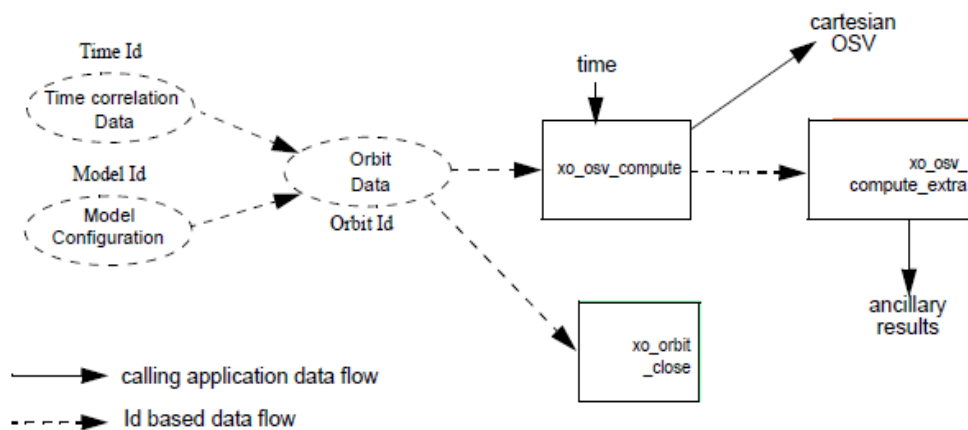


Figure 4: Propag Routines Data Flow

All the previous function are described in [ORB_SUM].

Example 4.14 - I: Orbit computation

```

/* Variables */
long  status, func_id, n;
char  msg[XL_MAX_COD][XL_MAX_STR];
long  xl_ierr[XL_ERR_VECTOR_MAX_LENGTH];
long  xo_ierr[XO_ERR_VECTOR_MAX_LENGTH];

long      sat_id      = XO_SAT_CRYOSAT;
xl_time_id time_id    = {NULL};
xl_model_id model_id  = {NULL};
xo_orbit_id orbit_id  = {NULL};

double tri_time[4];
double tri_orbit_num, tri_anx_time, tri_orbit_duration;

long time_ref;
double time;
double pos_ini[3], vel_ini[3],
       pos[3],    vel[3];
xo_validity_time val_times;
double val_time0, val_time1;
long abs_orbit;
    
```

Variable declaration

```

/* Time initialisation */
tri_time[0] = -245.100000000; /* TAI time [days] */
tri_time[1] = tri_time[0] - 35.0/86400.; /* UTC time [days] (= TAI - 35.0 s) */
tri_time[2] = tri_time[0] - 35.3/86400.; /* UT1 time [days] (= TAI - 35.3 s) */
tri_time[3] = tri_time[0] - 19.0/86400.; /* GPS time [days] (= TAI - 19.0 s) */

tri_orbit_num = 10;
tri_anx_time  = 5245.123456;
tri_orbit_duration = 6035.928144;

status = xl_time_ref_init(tri_time, &tri_orbit_num, &tri_anx_time,
                          &tri_orbit_duration, &time_id, xl_ierr);
if (status != XL_OK)
{
    func_id = XL_TIME_REF_INIT_ID;
    xl_get_msg(&func_id, xl_ierr, &n, msg);
    xl_print_msg(&n, msg);
    if (status <= XL_ERR) return(XL_ERR);
}
    
```

Time Initialisation

```

/* Orbit initialisation */
time_ref = XL_TIME_UT1;
time = -2452.569;
pos_ini[0] = 6427293.5314;
pos_ini[1] = -3019463.3246;
pos_ini[2] = 0;

vel_ini[0] = -681.1285;
vel_ini[1] = -1449.8649;
vel_ini[2] = 7419.5081;

status = xo_orbit_cart_init(&sat_id, &model_id, &time_id,
                           &time_ref, &time,
                           pos_ini, vel_ini, &abs_orbit,
                           &val_time0, &val_time1, &orbit_id,
                           xo_ierr);

if (status != XO_OK)
{
  func_id = XO_ORBIT_CART_INIT_ID;
  xo_get_msg(&func_id, ierr, &n, msg);
  xo_print_msg(&n, msg);
}

```

Orbit Initialisation

```

/* Get propagation validity interval*/
init_mode = XO_SEL_DEFAULT; /* select the default time */

status = xo_orbit_get_osv_compute_validity(&orbit_id, &val_times);
if (status == XO_OK)
{
  printf("\t- Propagation validity times = ( %lf , %lf )\n",
        val_times.start, val_times.stop );
}

```

Get propagation validity times

```

/* propagation: loop to propagate along the validity interval */
time_ref = val_times.time_ref;
for ( time = val_times.start;
      time < val_times.stop;
      time += ((val_time1-val_time0)/10) )
{
  status = xo_osv_compute(&orbit_id, &propag_model, &time_ref, &time,
                          pos, vel, acc, xo_ierr);

  if (status != XO_OK)
  {
    func_id = XO_OSV_COMPUTE_ID;
    xo_get_msg(&func_id, ierr, &n, msg);
    xo_print_msg(&n, msg);
  }

  printf("\t- Time           = %lf\n", time );
  printf("\t- Position        = (%lf, %lf, %lf)\n", pos[0], pos[1], pos[2]);
  printf("\t- Velocity         = (%lf, %lf, %lf)\n", vel[0], vel[1], vel[2]);
  printf("\t- Acceleration     = (%lf, %lf, %lf)\n", acc[0], acc[1], acc[2]);
}

```

Orbit State Vector computation

```
/* Close orbit_id */  
status = xo_orbit_close(&orbit_id, xo_ierr);  
if (status != XO_OK)  
{  
    func_id = XO_ORBIT_CLOSE_ID;  
    xo_get_msg(&func_id, xo_ierr, &n, msg);  
    xo_print_msg(&n, msg);  
}
```

Orbit close

```
/* Close time reference */  
status = xl_time_close(&time_id, xl_ierr);  
if (status != XO_OK)  
{  
    func_id = XL_TIME_CLOSE_ID;  
    xo_get_msg(&func_id, xl_ierr, &n, msg);  
    xo_print_msg(&n, msg);  
}
```

Time Close

4.16 Generation of Earth Observation Orbit Mission Files

The Earth Observation files allow the generation of different orbit files types:

- Orbit Scenario files: **xo_gen_osf_create**.
- Predicted Orbit files: **xo_gen_pof**
- Restituted Orbit files (DORIS restituted and DORIS precise): **xo_gen_rof**
- DORIS Navigator files: **xo_gen_dnf**
- Orbit Event files: **xo_gen_oef**.

The strategy to follow in all cases is similar:

- Initialise the time correlations (see section 4.8) to create the *xl_time_id* that will be used in the generation functions. This step is not needed for the generation of orbit event files.
- Call one of the generation function described above.
- Optionally for the generation of orbit scenario files: it is possible to add orbital changes within the orbit scenario file by calling one of this functions: **xo_gen_osf_append_orbit_change**, **xo_gen_osf_repeath_cycle**, **xo_gen_osf_add_drift_cycle**.
- Close time correlations (see section 4.8). This step is not needed for the generation of orbit event files.

Additionally there exists a set of executable programs that are equivalent to the previous functions.

More information can be found in [ORB_SUM].

Next figure shows the calling sequence for the file generation functions.

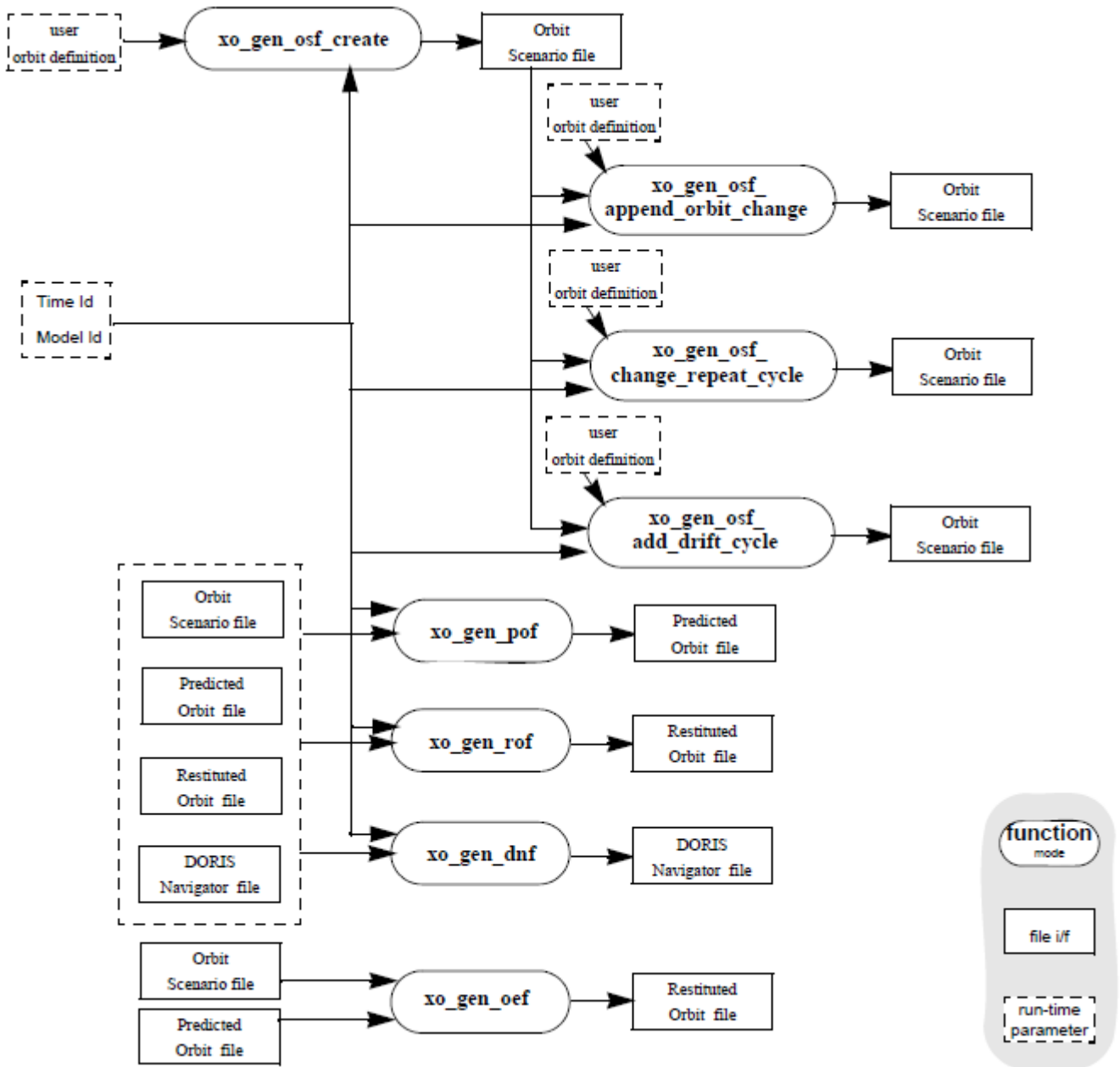


Figure 5: File Generation Calling Sequence

Example 4.15 - I: Orbit Scenario file generation

```

/* Variables */
long   status, func_id, n;
char   msg[XL_MAX_COD][XL_MAX_STR];
long   xl_ierr[XL_ERR_VECTOR_MAX_LENGTH];
long   xo_ierr[XO_ERR_VECTOR_MAX_LENGTH];

long   sat_id           = XO_SAT_CRYOSAT;
xl_time_id time_id     = {NULL};
xl_model_id model_id   = {NULL};

double tri_time[4];
double tri_orbit_num, tri_anx_time, tri_orbit_duration;

long   abs_orbit,   cycle, phase, phase_inc;
long   repeat_cycle,   cycle_length;
long   drift_mode;
double mlst_drift, mlst,   anx_long;
long   osf_version = 1;
char   file_class[] = "TEST";
char   fh_system = "CFI Example";

char   output_dir[] = "";
char   output_file_1[] = "my_osf.eef" /* name for the output osf */
char   output_file_2[] = "osf_after_append.eef" /* name for the output osf */
    
```

Variable declaration

```

/* Time initialisation */

tri_time[0] = -245.100000000; /* TAI time [days] */
tri_time[1] = tri_time[0] - 35.0/86400.; /* UTC time [days] (= TAI - 35.0 s) */
tri_time[2] = tri_time[0] - 35.3/86400.; /* UT1 time [days] (= TAI - 35.3 s) */
tri_time[3] = tri_time[0] - 19.0/86400.; /* GPS time [days] (= TAI - 19.0 s) */

tri_orbit_num = 10;
tri_anx_time = 5245.123456;
tri_orbit_duration = 6035.928144;

status = xl_time_ref_init(tri_time, &tri_orbit_num, &tri_anx_time,
                          &tri_orbit_duration, &time_id, xl_ierr);
if (status != XL_OK)
{
    func_id = XL_TIME_REF_INIT_ID;
    xl_get_msg(&func_id, xl_ierr, &n, msg);
    xl_print_msg(&n, msg);
    if (status <= XL_ERR) return(XL_ERR);
}
    
```

Time Initialisation


```

/* Generate the OSF */

date = 1643.39513888889; /* UTC=2004-07-01_09:29:00.000000 */
abs_orbit = 1;
cycle = 1;
phase = 1;
repeat_cycle = 369;
cycle_length = 5344;
drift_mode = XO_NOSUNSYNC_DRIFT;
mlst_drift = -179.208556;
mlst = 12.0;
anx_long = 37.684960;
osf_version = 1;

status = xo_gen_osf_create(&sat_id, &model_id, &time_id, &abs_orbit,
                           &cycle, &phase, &repeat_cycle, &cycle_length,
                           &anx_long, &drift_mode, &inclination,
                           &mlst_drift, &mlst, &date,
                           output_dir, output_file,
                           file_class, &osf_version, fh_system,
                           xo_ierr);

if (status != XO_OK)
{
  func_id = XO_GEN_OSF_CREATE_ID;
  xo_get_msg(&func_id, xo_ierr, &n, msg);
  xo_print_msg(&n, msg);
}

```

```

/* Append an orbital change to the generated OSF */

old_nodal_period = 86400.0*(1+mlst_drift/86400.0)*
                  (double)repeat_cycle/(double)cycle_length;
osf_version++;
abs_orbit = 5345;
phase_inc = XO_NO_PHASE_INCREMENT;
repeat_cycle = 369;
cycle_length = 5344;

/* small change wrt to nominal to check tolerances */
anx_long = 37.68497;
mlst = mlst + mlst_drift*(5345-1)*old_nodal_period/(3600.0*86400.0) + 24.0;

status = xo_gen_osf_append_orbit_change(&sat_id, &model_id, &time_id,
                                         output_file_1, &abs_orbit,
                                         &repeat_cycle, &cycle_length,
                                         &anx_long, &drift_mode,
                                         &inclination, &mlst_drift,
                                         &mlst, &phase_inc,
                                         output_dir, output_file_2,
                                         file_class, &osf_version, fh_system,
                                         xo_ierr);

if (status != XO_OK)
{
  func_id = XO_GEN_OSF_APPEND_ORBIT_CHANGE_ID;
  xo_get_msg(&func_id, ierr, &n, msg);
  xo_print_msg(&n, msg);
}

```

```

/* Close time references */
status = xl_time_close(&time_id, xl_ierr);
[...]
    
```

Time
close

Example 4.15 - II: Predicted Orbit file generation

```

/* Variables */
long    status, func_id, n;
char    msg[XL_MAX_COD][XL_MAX_STR];
long    xl_ierr[XL_ERR_VECTOR_MAX_LENGTH];
long    xo_ierr[XO_ERR_VECTOR_MAX_LENGTH];

long    sat_id    = XO_SAT_CRYOSAT;
xl_model_id model_id = {NULL};
xl_time_id time_id = {NULL};

double  tri_time[4];
double  tri_orbit_num, tri_anx_time, tri_orbit_duration;

char    reference_file[] = "input_osf_file";
char    pof_filename[XD_MAX_STRING] = "";
char    output_directory[XD_MAX_STRING] = "";

long    time_mode,    time_ref;
double  start_time, stop_time;
double  osv_location;
long    ref_filetype;

char    file_class[] = "TEST";
long    version_number = 1;
char    fh_system = "CFI Example";
    
```

Variable declaration

```

/* Time initialisation */
tri_time[0] = -245.100000000; /* TAI time [days] */
tri_time[1] = tri_time[0] - 35.0/86400.; /* UTC time [days] (= TAI - 35.0 s) */
tri_time[2] = tri_time[0] - 35.3/86400.; /* UT1 time [days] (= TAI - 35.3 s) */
tri_time[3] = tri_time[0] - 19.0/86400.; /* GPS time [days] (= TAI - 19.0 s) */

tri_orbit_num = 10;
tri_anx_time = 5245.123456;
tri_orbit_duration = 6035.928144;

status = xl_time_ref_init(tri_time, &tri_orbit_num, &tri_anx_time,
                          &tri_orbit_duration, &time_id, xl_ierr);
if (status != XL_OK)
{
    func_id = XL_TIME_REF_INIT_ID;
    xl_get_msg(&func_id, xl_ierr, &n, msg);
    xl_print_msg(&n, msg);
    if (status <= XL_ERR) return(XL_ERR);
}
    
```

Time Initialisation

```

/* Generate the POF */
time_mode = XO_SEL_TIME;
time_ref = XO_TIME_UTC;
start_time = 1646.0;
stop_time = 1647.0;
osv_location = 0.0;
ref_filetype = XO_REF_FILETYPE_OSF;

status = xo_gen_pof(&sat_id, &model_id, &time_id,
                  &time_mode, &time_ref, &start_time,
                  &stop_time, &start_orbit, &stop_orbit,
                  &osv_location, &ref_filetype,
                  reference_file, &pof_filetype, output_directory,
                  pof_filename, file_class, &version_number, fh_system,
                  xo_ierr);

if (status != XO_OK)
{
    func_id = XO_GEN_POF_ID;
    xo_get_msg(&func_id, xo_ierr, &n, msg);
    xo_print_msg(&n, msg);
}
    
```

Generate Predicted Orbit File

```

/* Close time references */
status = xl_time_close(&time_id, xl_ierr);
[...]
    
```

Time close

Example 4.15 - III: Restituted Orbit file generation

```

/* Variables */
long status, func_id, n;
char msg[XL_MAX_COD][XL_MAX_STR];
long xl_ierr[XL_ERR_VECTOR_MAX_LENGTH];
long xo_ierr[XO_ERR_VECTOR_MAX_LENGTH];
long sat_id = XO_SAT_CRYOSAT;
xl_model_id model_id = {NULL};
xl_time_id time_id = {NULL};
double tri_time[4], tri_orbit_num, tri_anx_time, tri_orbit_duration;
char reference_file[] = "input_osf_file";
char rof_filename[XD_MAX_STRING] = "";
char output_directory[XD_MAX_STRING] = "";
long time_mode, time_ref;
double start_time, stop_time, osv_interval;
long ref_filetype, osv_precise, rof_filetype;
char file_class[] = "TEST";
long version_number = 1;
char fh_system = "CFI Example";
    
```

Variable declaration

```

/* Time initialisation */
tri_time[0] = -245.100000000;          /* TAI time [days] */
tri_time[1] = tri_time[0] - 35.0/86400.; /* UTC time [days] (= TAI - 35.0 s) */
tri_time[2] = tri_time[0] - 35.3/86400.; /* UT1 time [days] (= TAI - 35.3 s) */
tri_time[3] = tri_time[0] - 19.0/86400.; /* GPS time [days] (= TAI - 19.0 s) */
tri_orbit_num = 10;
tri_anx_time = 5245.123456;
tri_orbit_duration = 6035.928144;

status = xl_time_ref_init(tri_time, &tri_orbit_num, &tri_anx_time,
                          &tri_orbit_duration, &time_id, xl_ierr);
if (status != XL_OK)
{
  func_id = XL_TIME_REF_INIT_ID;
  xl_get_msg(&func_id, xl_ierr, &n, msg);
  xl_print_msg(&n, msg);
  if (status <= XL_ERR) return(XL_ERR);
}

```

Time Initialisation

```

/* Generate the ROF */
time_mode = XO_SEL_TIME;
time_ref = XO_TIME_UTC;
start_time = 1646.0;
stop_time = 1646.2;
osv_interval = 60;
osv_precise = XO_OSV_PRECISE_MINUTE;
ref_filetype = XO_REF_FILETYPE_OSF;
rof_filetype = XO_REF_FILETYPE_ROF;

status = xo_gen_rof(&sat_id, &model_id, &time_id,
                   &time_mode, &time_ref, &start_time,
                   &stop_time, &start_orbit, &stop_orbit,
                   &osv_interval, &osv_precise, &ref_filetype,
                   reference_file, &rof_filetype, output_directory,
                   rof_filename, file_class, &version_number, fh_system,
                   xo_ierr);
if (status != XO_OK)
{
  func_id = XO_GEN_ROF_ID;
  xo_get_msg(&func_id, xo_ierr, &n, msg);
  xo_print_msg(&n, msg);
}

```

Generate Restituted Orbit File

```

/* Close time references */
status = xl_time_close(&time_id, xl_ierr);
[...]

```

Time close

Example 4.15 - IV: Executable program for generating a Restituted orbit file

The following command line generates the same file that the code in Example 4.15 - III

```

gen_rof -sat CRYOSAT -tref UTC -tstart 1646.0 -tstop 1646.2 -osvint 60 \
        -reftyp OSF -ref input_osf_file \
        -roftyp ROF -rof ROF_example_file.EEF \
        -tai 0.0000 -gps 0.00021991 -utc 0.00040509 -ut1 0.00040865

```

Example 4.15 - V: Restituted Orbit file generation

```
/* Variables */  
long  status, func_id, n;  
char  msg[XL_MAX_COD][XL_MAX_STR];  
long  xo_ierr[XO_ERR_VECTOR_MAX_LENGTH];  
  
char  file_class[] = "TEST";  
long  version_number = 1;  
char  fh_system = "CFI Example";  
  
char  oef_filename[XD_MAX_STR];  
char  osf_filename[] = "input_osf.eef";  
char  pof_filename[] = "input_pof.eef";
```

Variable declaration

```
/* Generate the OEF */  
status = xo_gen_oef(oef_filename, osf_filename, pof_filename,  
                  file_class, &version_number, fh_system,  
                  xo_ierr);  
if (status != XO_OK)  
{  
    func_id = XO_GEN_OEF_ID;  
    xo_get_msg(&func_id, xo_ierr, &n, msg);  
    xo_print_msg(&n, msg);  
}
```

OEF generation

4.17 Target calculation

This functionality allows to perform accurate computation of pointing parameters from and to a satellite for various types of targets.

Before the user could call targets function, some parameters has to be initialised:

- Attitude: The attitude defines the relation between coordinate frames related to the satellite and a general reference frame. In order to define the attitude, the user has to call some initialisation functions that generate another CFI Id called *xp_attitude_id*. (See section 4.17.1 for further details about attitude initialisation)
- For some targets calculation it could be needed to take into account the atmospheric refraction of a signal travelling to/from the satellite. In these cases the user could choose the atmospheric model to use. For using an atmospheric model in the target calculation, a CFI Id called *xp_atmos_id* has to be initialised previously, afterwards it is introduced in the target functions.(See section 4.17.2 for further details about atmospheric initialisation)
- For geolocation routines it could be needed a digital elevation model (DEM) in order to provide a more accurate target. The DEM is introduced in the target calculation using the CFI Id structured called *xp_dem_id*. This Id has to be initialised previously to the target calculation.(See section 4.17.3 for further details about DEM initialisation)

4.17.1 Attitude initialisation

The initialisation strategy for the attitude is the following:

- Satellite and instrument attitude frames initialisation. There are three different levels of attitude frames defined for this issue (see [MCD]):
 - Satellite Nominal Attitude Frame.
 - Satellite Attitude Frame
 - Instrument Attitude Frame

Each of the frames is defined independently and produce a CFI Id where the initialisation parameters are stored. Note that not all attitude frames has to be defined. There are a set of functions to initialise each frame depending on the type of parameters used to establish the reference frame (see Figure 6, Figure 7 and Figure 8)

- Attitude initialisation. Using the function **xp_attitude_init**, the CFI Id *xp_attitude_id* is initialised. At this stage, the structure doesn't contain attitude data and it cannot be used in target functions.
- Attitude computation: Using a satellite state vector at a given time and the attitude frames previously initialised, the *xp_attitude_id* structure is filled in. by calling the function **xp_attitude_compute**.

All functions for attitude computation are explained in detail in [PNT_SUM].

The typical data flow for the attitude functions described above is shown schematically in the Figure 9.

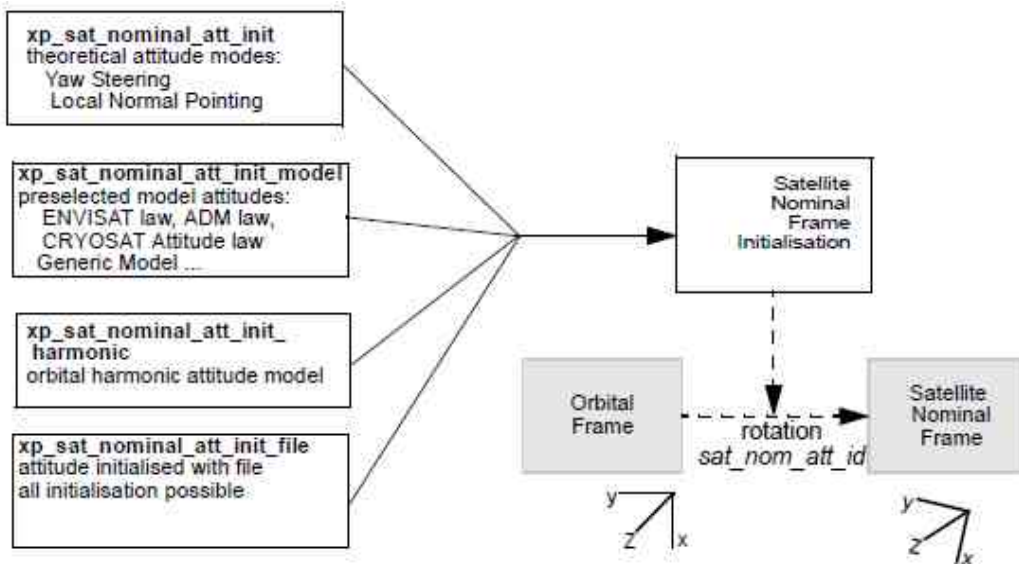


Figure 6: Satellite Nominal Initialisation

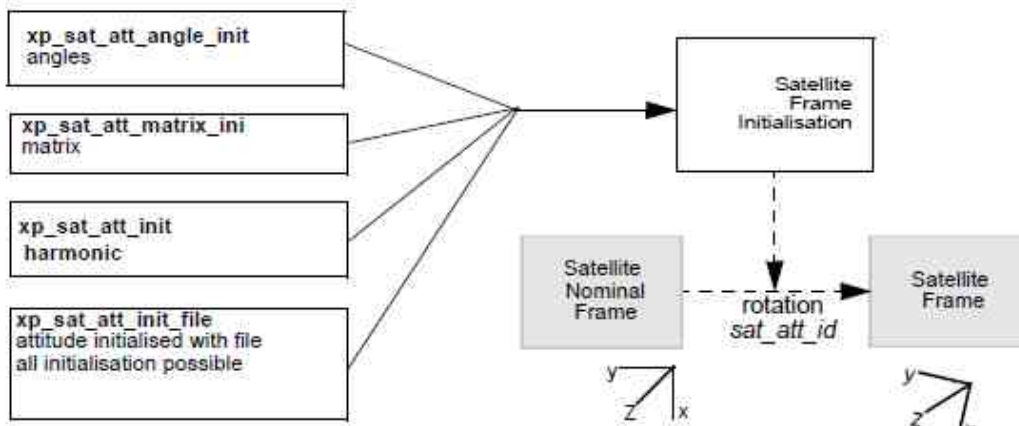


Figure 7: Satellite Initialisation

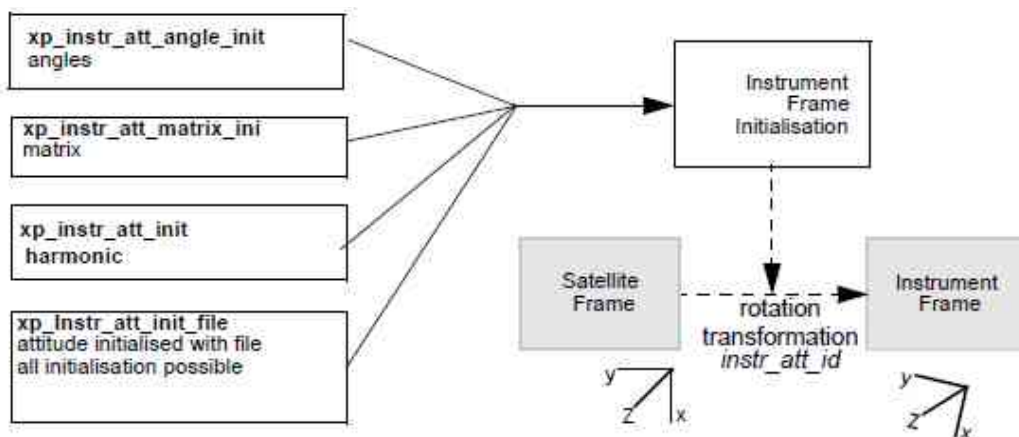


Figure 8: Instrument Initialisation

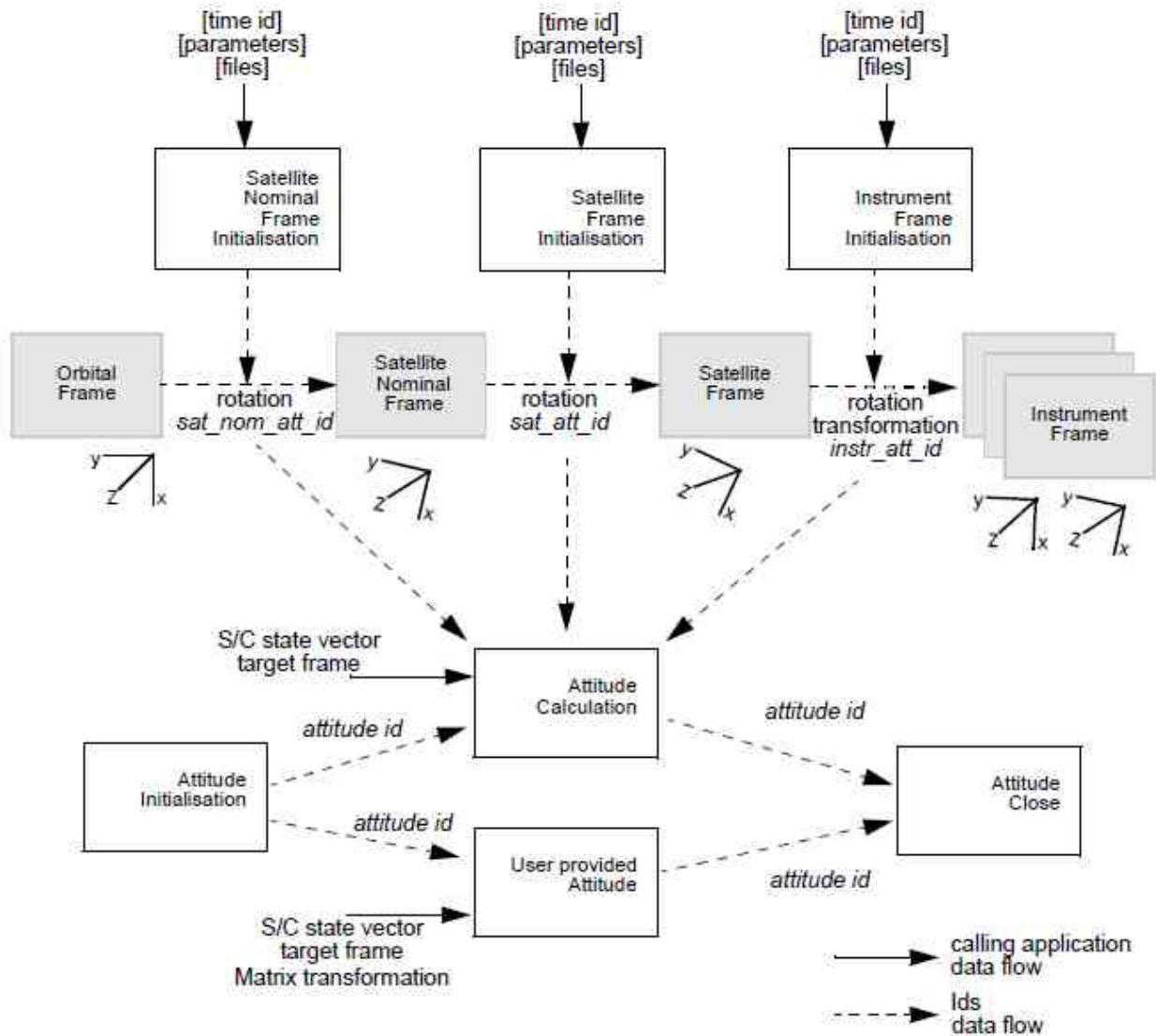


Figure 9: Attitude data flow

Example 4.16 - I: ENVISAT AOCS model plus mispointing angles

```

/* Variables */
long   status, func_id, n;
char   msg[XL_MAX_COD][XL_MAX_STR];
long   xl_ierr[XL_ERR_VECTOR_MAX_LENGTH];
long   xp_ierr[XP_ERR_VECTOR_MAX_LENGTH];

long   sat_id           = XO_SAT_ENVISAT;
xl_model_id   model_id = {NULL};
xl_time_id    time_id   = {NULL};
xp_sat_nom_trans_id sat_nom_trans_id = {NULL};
xp_sat_trans_id sat_trans_id = {NULL};
xp_instr_trans_id instr_trans_id = {NULL};
xp_attitude_id attitude_id = {NULL};

double tri_time[4];
double tri_orbit_num, tri_anx_time, tri_orbit_duration;

long model_enum;
double model_param[XP_NUM_MODEL_PARAM];
double ang[3];
xp_param_model_str param_model;

long time_ref;
double time;
double pos[3], vel[3], acc[3];
    
```

Variable declaration

```

/* Time initialisation */
tri_time[0] = -245.100000000; /* TAI time [days] */
tri_time[1] = tri_time[0] - 35.0/86400.; /* UTC time [days] (= TAI - 35.0 s) */
tri_time[2] = tri_time[0] - 35.3/86400.; /* UT1 time [days] (= TAI - 35.3 s) */
tri_time[3] = tri_time[0] - 19.0/86400.; /* GPS time [days] (= TAI - 19.0 s) */

tri_orbit_num = 10;
tri_anx_time = 5245.123456;
tri_orbit_duration = 6035.928144;

status = xl_time_ref_init(tri_time, &tri_orbit_num, &tri_anx_time,
                        &tri_orbit_duration, &time_id, xl_ierr);
if (status != XL_OK)
{
    func_id = XL_TIME_REF_INIT_ID;
    xl_get_msg(&func_id, xl_ierr, &n, msg);
    xl_print_msg(&n, msg);
    if (status <= XL_ERR) return(XL_ERR);
}
    
```

Time Initialisations

```
/* Satellite Nominal Attitude frame initialisation */

model_enum = XP_MODEL_ENVISAT;
model_param[0] = -0.1671;
model_param[1] = 0.0501;
model_param[2] = 3.9130;

local_status = xp_sat_nominal_att_init_model(&model_enum, model_param,
                                             &sat_nom_trans_id, xp_ierr);

if (status != XP_OK)
{
    func_id = XP_SAT_NOMINAL_ATT_INIT_MODEL_ID;
    xp_get_msg(&func_id, xp_ierr, &n, msg);
    xp_print_msg(&n, msg);
    if (status <= XP_ERR) return(XP_ERR);
}
}
```

Satellite Nominal Attitude frame

```
/* Satellite Attitude frame initialisation */
ang[0] = 0.0046941352;
ang[1] = 0.0007037683;
ang[2] = 356.09346792;

local_status = xp_sat_att_angle_init(ang, &sat_trans_id, xp_ierr);

if (status != XP_OK)
{
    func_id = XP_SAT_ATT_ANGLE_INIT_ID;
    xp_get_msg(&func_id, xp_ierr, &n, msg);
    xp_print_msg(&n, msg);
    if (status <= XP_ERR) return(XP_ERR);
}
}
```

Satellite Attitude frame

```
/* attitude initialisation */

status = xp_attitude_init (&attitude_id, xp_ierr);

if (status != XL_OK)
{
    func_id = XP_ATTITUDE_INIT_ID;
    xp_get_msg(&func_id, xp_ierr, &n, msg);
    xp_print_msg(&n, msg);
}
}
```

Attitude initialisation

```

/* Get attitude */
target_frame = XP_SAT_ATT;
time_ref = XL_TIME_UTC; /* Satellite state vector */
time      = 255.3456;
pos[0]    = 6997887.57;
pos[1]    = -1536046.83;
pos[2]    = 99534.18;
vel[0]    = -240.99;
vel[1]    = -1616.85;
vel[2]    = -7376.65;
acc[0]    = -7.79104;
acc[1]    = 1.69353;
acc[2]    = -0.10826;
local_status = xp_attitude_compute(&model_id, &time_id, &sat_nom_trans_id,
                                   &sat_trans_id, &instr_trans_id,
                                   &attitude_id, &time_ref, &time,
                                   pos, vel, acc, &target_frame, xp_ierr);

if (status != XP_OK)
{
  func_id = XP_ATTITUDE_COMPUTE_ID;
  xp_get_msg(&func_id, xp_ierr, &n, msg);
  xp_print_msg(&n, msg);
  if (status <= XP_ERR) return(XP_ERR);
}

```

Attitude computation

```

/* Get attitude data */
status = xp_attitude_get_id_data(&attitude_id, &attitude_data);
printf("- Init Status      : %li\n", xp_attitude_init_status(&attitude_id));
printf("- Init Mode        : %li\n", xp_attitude_get_mode(&attitude_id));
printf("- Model            : %li\n", attitude_data.model);
printf("- Time Reference    : %li\n", attitude_data.time_ref);
printf("- Time              : %lf\n", attitude_data.time);
printf("- Sat Position      : [%12.3lf,%12.3lf,%12.3lf]\n",
      attitude_data.sat_vector.v[0],
      attitude_data.sat_vector.v[1],
      attitude_data.sat_vector.v[2]);

[...]
printf("- Source frame      : %lf\n", attitude_data.source_frame);
printf("- Target frame     : %lf\n", attitude_data.target_frame);
printf("- Attitude Matrix   : %lf\t%lf%lf\n",
      attitude_data.sat_mat.m[0][0], attitude_data.sat_mat.m[0][1],
      attitude_data.sat_mat.m[0][2]);
printf("
      %lf\t%lf%lf\n",
      attitude_data.sat_mat.m[1][0], attitude_data.sat_mat.m[1][1],
      attitude_data.sat_mat.m[1][2]);
printf("
      %lf\t%lf%lf\n",
      attitude_data.sat_mat.m[2][0], attitude_data.sat_mat.m[2][1],
      attitude_data.sat_mat.m[2][2]);

[...]

```

Getting attitude data

```

/* Get the attitude for a new satellite position
   Note that it is not necessary to close the attitude_id */
target_frame = XP_SAT_ATT;
time_ref = XL_TIME_UTC; /* Satellite state vector */
time      = 255.3456;
pos[0]    = 4859964.138;
pos[1]    = -5265612.059;
pos[2]    = 0.002;
vel[0]    = -1203.303801;
vel[1]    = -1098.845511;
vel[2]    = 7377.224410;
acc[0]    = 0.0;
acc[1]    = 0.0;
acc[2]    = 0.0;
local_status = xp_attitude_compute(&model_id, &time_id,
                                   &sat_nom_trans_id,
                                   &sat_trans_id, &instr_trans_id,
                                   &attitude_id, &time_ref, &time,
                                   pos, vel, acc, &target_frame,
                                   xp_ierr);

if (status != XP_OK)
{
    func_id = XP_ATTITUDE_COMPUTE_ID;
    xp_get_msg(&func_id, xp_ierr, &n, msg);
    xp_print_msg(&n, msg);
    if (status <= XP_ERR) return(XP_ERR);
}
    
```

Attitude computation

```

/* Close attitude */
status = xp_attitude_close(&attitude_id, xp_ierr);
if (status != XL_OK)
{
    func_id = XP_ATTITUDE_CLOSE_ID;
    xp_get_msg(&func_id, xp_ierr, &n, msg);
    xp_print_msg(&n, msg);
}
    
```

Attitude close

```

/* Close Satellite Nominal Attitude frame */
status = xp_sat_nominal_att_close(&sat_nom_trans_id, xp_ierr);
if (status != XL_OK)
{
    func_id = XP_SAT_NOMINAL_ATT_CLOSE_ID;
    xp_get_msg(&func_id, xp_ierr, &n, msg);
    xp_print_msg(&n, msg);
}
    
```

Close Sat Att. Nom.

```

/* Close Satellite Attitude frame */
status = xp_sat_att_close(&sat_trans_id, xp_ierr);
if (status != XL_OK)
{
    func_id = XP_SAT_ATT_CLOSE_ID;
    xp_get_msg(&func_id, xp_ierr, &n, msg);
    xp_print_msg(&n, msg);
}
    
```

Close Sat Att. frame

```

/* Close time_id */
status = xp_time_close(&time_id, xl_ierr);
if (status != XL_OK)
{
    func_id = XP_TIME_CLOSE_ID;
    xp_get_msg(&func_id, xl_ierr, &n, msg);
    xp_print_msg(&n, msg);
}
    
```

Time close

Example 4.16 - II: Attitude defined by star tracker for cryosat

```

/* Variables */
[...]
char att_file[] = "../..data/CRYOSAT_STAR_TRACKER_DATA.DBL";
char auxiliary_file[] = "../..data/cryosat_reference_frame_conf.xml";

[ ... Time initialisation... ]
    
```

```

/* satellite reference initialization */

files[0] = att_file;
n_files = 1;
time_init_mode = XO_SEL_FILE;
time_ref = XL_TIME_UTC;
time0 = 1646.50;
time1 = 1646.60;
target_frame = XP_SAT_ATT;

status = xp_sat_att_init_file(&time_id, &n_files, files, auxiliary_file,
                             &time_init_mode, &time_ref, &time0, &time1,
                             &val_time0, &val_time1,
                             &sat_trans_id, xp_ierr);

if (status != XL_OK)
{
    func_id = XP_SAT_ATT_INIT_FILE_ID;
    xp_get_msg(&func_id, xp_ierr, &n, msg);
    xp_print_msg(&n, msg);
}
    
```

Satellite Attitude frame

```

/* attitude initialisation */
status = xp_attitude_init (&attitude_id, xp_ierr);
if (status != XL_OK)
{
    func_id = XP_ATTITUDE_INIT_ID;
    xp_get_msg(&func_id, xp_ierr, &n, msg);
    xp_print_msg(&n, msg);
}
    
```

Attitude Initialisation

```
/* attitude computation */
time = 1646.775;
pos[0] = +2117636.668;
pos[1] = -553780.175;
pos[2] = -6748229.578;
vel[0] = +6594.65340;
vel[1] = -2760.52030;
vel[2] = +2303.10280;

status = xp_attitude_compute(&model_id, &time_id,
                             &sat_nom_trans_id,
                             &sat_trans_id, &instr_trans_id,
                             &attitude_id, &time_ref, &time,
                             pos, vel, acc, &target_frame,
                             xp_ierr);

if (status != XL_OK)
{
    func_id = XP_ATTITUDE_COMPUTE_ID;
    xp_get_msg(&func_id, xp_ierr, &n, msg);
    xp_print_msg(&n, msg);
}
}
```

Attitude computation

[... Attitude usage...]

```
/* Close attitude */
status = xp_attitude_close(&attitude_id, xp_ierr);
if (status != XL_OK)
{
    func_id = XP_ATTITUDE_CLOSE_ID;
    xp_get_msg(&func_id, xp_ierr, &n, msg);
    xp_print_msg(&n, msg);
}
}
```

Attitude close

```
/* Close Satellite Attitude frame */
status = xp_sat_att_close(&sat_trans_id, xp_ierr);
if (status != XL_OK)
{
    func_id = XP_SAT_ATT_CLOSE_ID;
    xp_get_msg(&func_id, xp_ierr, &n, msg);
    xp_print_msg(&n, msg);
}
}
```

Attitude frame close

[Close time_id ...]

4.17.2 Atmospheric initialisation

When using an atmospheric model, the ID *xp_atmos_id* structure should be initialised by calling the CFI function **xp_atmos_init** (see [PNT_SUM]) providing the needed atmospheric model and files.

Once the *xp_atmos_id* has been initialised, it can be used as an input parameter for target calculations (see section 4.17.4).

The memory allocated for *xp_atmos_id* should be freed when the structure is not to be used in the program by calling the CFI function **xp_atmos_close**.

4.17.3 Digital Elevation model

Before using a digital elevation model, the ID *xp_dem_id* structure should be initialised by calling the CFI function **xp_dem_init** (see [PNT_SUM]) providing the configuration file for the DEM.

Once the *xp_dem_id* has been initialised, it can be used as an input parameter for target calculations (see section 4.17.4).

The memory allocated for *xp_dem_id* should be freed when the structure is not to be used in the program by calling the CFI function **xp_dem_close**.

4.17.4 Targets

Once the attitude has been initialised and optionally have the atmospheric and the DEM models, the targets can be calculated. For this issue there is a set of functions that solves different types of pointing problems. A detailed explanation of the different target problems can be seen in [PNT_SUM] section 4.

For every target problem, three different target types are defined:

- User target: it is the target requested by the user.
- LOS target (line of sight target): it is the computed raypath to reach the user target.
- DEM target: it is a target computed taking into account the DEM model. It is only used for geolocated targets.

The previous functions do not return directly the computed target parameters, but another ID called *xp_target_id*. The target data for one of the target types (user, LOS or DEM) has to be retrieved from the *xp_target_id* using another set of functions called **xp_target_extra_xxx**.

Once a target is not to be used any more, it has to be closed in order to free internal memory by calling **xp_target_close**.

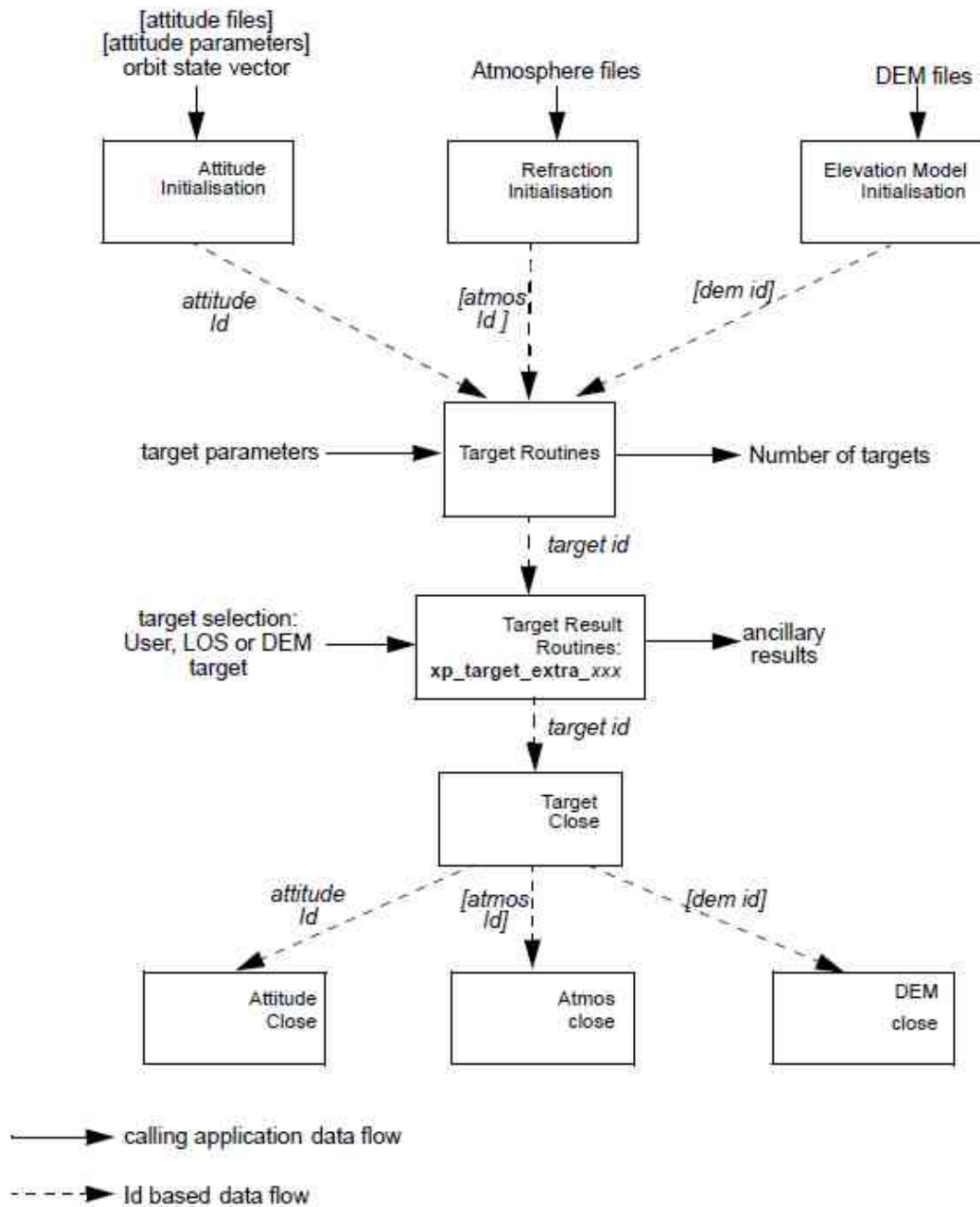


Figure 10: Target data flow

The following figure summarises the data flow for the target calculation:

Example 4.16 - III: Target Star

```
/* Local Variables */
[...]  

[ ... Time initialisation...]
```

```
/* Satellite Nominal attitude frame initialisation */  

sat_id      = XP_SAT_ENVISAT;  

model_enum  = XP_MODEL_ENVISAT;  

model_param[0] = -0.1671;  

model_param[1] = 0.0501;  

model_param[2] = 3.9130;  
  

local_status = xp_sat_nominal_att_init_model(&model_enum, model_param,  

                                             &sat_nom_trans_id, xp_ierr);  

if (status != XP_OK)  

{  

  func_id = XP_SAT_NOMINAL_ATT_INIT_MODEL_ID;  

  xp_get_msg(&func_id, xp_ierr, &n, msg);  

  xp_print_msg(&n, msg);  

  if (status <= XP_ERR) return(XP_ERR);  

}
```

Satellite Nominal
Attitude frame

```
/* Attitude initialisation */  

status = xp_attitude_init (&attitude_id, xp_ierr);  

if (status != XP_OK)  

{  

  func_id = XP_ATTITUDE_INIT_ID;  

  xp_get_msg(&func_id, xp_ierr, &n, msg);  

  xp_print_msg(&n, msg);  

  if (status <= XP_ERR) return(XP_ERR);  

}
```

Attitude
Initialisation

```
/* Attitude computation */  

time_ref    = XL_TIME_UT1;  

time        = 255.3456;  

pos[0]      = 4859964.138;  

pos[1]      = -5265612.059;  

pos[2]      = 0.002;  

vel[0]      = -1203.303801;  

vel[1]      = -1098.845511;  

vel[2]      = 7377.224410;  

acc[0]      = 0.0;  

acc[1]      = 0.0;  

acc[2]      = 0.0;  

target_frame = XP_SAT_NOM_ATT;  
  

status = xp_attitude_compute(&model_id, &time_id, &sat_nom_trans_id,  

                             &sat_trans_id, &instr_trans_id, &attitude_id,  

                             &time_ref, &time, pos, vel, acc,  

                             &target_frame, xp_ierr);  

if (status != XP_OK)  

{  

  func_id = XP_ATTITUDE_COMPUTE_ID;  

  xp_get_msg(&func_id, xp_ierr, &n, msg);  

  xp_print_msg(&n, msg);  

  if (status <= XP_ERR) return(XP_ERR);  

}
```

Attitude computations

```

/* Call xp_target_star function */

deriv      =      XL_DER_1ST;
star_ra    =      272.0;
star_dec   =      -73.0;
star_ra_rate =      0.0;
star_dec_rate =      0.0;
freq       =      1.e10;

status = xp_target_star(&sat_id, &attitude_id, &atmos_id, &dem_id,
                        &deriv, &star_ra, &star_dec,
                        &star_ra_rate, &star_dec_rate, &iray, &freq,
                        &num_user_target, &num_los_target,
                        &target_id, xp_ierr);

if (status != XP_OK)
{
  func_id = XP_TARGET_STAR_ID;
  xp_get_msg(&func_id, xp_ierr, &n, msg);
  xp_print_msg(&n, msg);
  if (status <= XP_ERR) return(XP_ERR);
}

```

```

/* Get user target parameters from the target_id */

choice      = XL_DER_1ST;
target_type = XP_USER_TARGET_TYPE;
target_number = 0;

status = xp_target_extra_vector(&target_id, &choice,
                                &target_type, &target_number,
                                results, results_rate,
                                results_rate_rate, xp_ierr);

if (status != XP_OK)
{
  func_id = XP_TARGET_EXTRA_VECTOR_ID;
  xp_get_msg(&func_id, xp_ierr, &n, msg);
  xp_print_msg(&n, msg);
  if (status <= XP_ERR) return(XP_ERR);
}

/* Print results */
printf(" OUTPUT \n");
printf("- Target Position : [%12.3lf,%12.3lf,%12.3lf]",
        results[0], results[1], results[2]);
printf("- Target Velocity : [%12.3lf,%12.3lf,%12.3lf]",
        results_rate[0], results_rate[1], results_rate[2]);
printf("- Range : %lf", results[6]);
printf("- Range Rate : %lf", results_rate[6]);
printf("- Sat-Target LOS : [%12.9lf,%12.9lf,%12.9lf]",
        results[3], results[4], results[5]);
printf("- Sat-Tar LOS Rate : [%12.9lf,%12.9lf,%12.9lf]",
        results_rate[3], results_rate[4], results_rate[5]);
[...]

```

```
/* Closing Ids */  
status = xp_target_close(&target_id, xp_ierr);  
{  
    func_id = XP_TARGET_CLOSE_ID;  
    xp_get_msg(&func_id, xp_ierr, &n, msg);  
    xp_print_msg(&n, msg);  
}
```

Close target

```
status = xp_attitude_close(&attitude_id, xp_ierr);  
{  
    func_id = XP_ATTITUDE_CLOSE_ID;  
    xp_get_msg(&func_id, xp_ierr, &n, msg);  
    xp_print_msg(&n, msg);  
}
```

Close attitude

```
status = xp_sat_nominal_att_close(&sat_nom_trans_id, xp_ierr);  
{  
    func_id = XP_SAT_NOMINAL_ATT_CLOSE;  
    xp_get_msg(&func_id, xp_ierr, &n, msg);  
    xp_print_msg(&n, msg);  
}
```

Close Sat Nom.
Att.

[Close time initialisation...]

Example 4.16 - IV: Target intersection: target computation along one orbit

The following code shows a complete example for:

- time initialisation
- Orbit initialisation
- Attitude initialisation
- Getting the intersection target for different points along one orbit

[...]

```

/* Local variables declaration */
long   status;
long   n;
long   func_id;
char   msg[XL_MAX_COD][XL_MAX_STR];
long   xl_ierr[XL_ERR_VECTOR_MAX_LENGTH];
long   xo_ierr[XO_ERR_VECTOR_MAX_LENGTH];
long   xp_ierr[XP_ERR_VECTOR_MAX_LENGTH];

long   sat_id;
xl_model_id   model_id   = {NULL};
xl_time_id    time_id    = {NULL};
xo_orbit_id   orbit_id   = {NULL};
xo_propag_id  propag_id  = {NULL};

xp_sat_nom_trans_id  sat_nom_trans_id = {NULL};
xp_sat_trans_id      sat_trans_id     = {NULL};
xp_instr_trans_id    instr_trans_id   = {NULL};
xp_attitude_id       attitude_id      = {NULL};

xp_atmos_id          atmos_id         = {NULL};
xp_dem_id            dem_id           = {NULL};
xp_target_id         target_id        = {NULL};
    
```

Variable declaration

[...]

```

/* Time initialization */
time_model   = XL_TIMEMOD_FOS_PREDICTED;
n_files      = 1;
time_init_mode = XL_SEL_FILE;
time_ref     = XL_TIME.UTC;
time0        = 0;
time1        = 0;
orbit0       = 0;
orbit1       = 0;
time_file[0] = orbit_file;

status = xl_time_ref_init_file(&time_model, &n_files, time_file,
                               &time_init_mode, &time_ref, &time0, &time1,
                               &orbit0, &orbit1, &val_time0, &val_time1,
                               &time_id, xl_ierr);

if (status != XL_OK)
{
    func_id = XL_TIME_REF_INIT_FILE_ID;
    xl_get_msg(&func_id, xo_ierr, &n, msg);
    xl_print_msg(&n, msg);
    if (status <= XL_ERR) return(XL_ERR);
}
    
```

Time Initialisation

```

/* Orbit initialization */
time_init_mode = XO_SEL_FILE;

input_files[0] = orbit_file;
n_files = 1;
orbit_mode = XO_ORBIT_INIT_AUTO;

status = xo_orbit_init_file(&sat_id, &model_id, &time_id,
                           &orbit_mode, &n_files, input_files,
                           &time_init_mode, &time_ref_utc,
                           &time0, &time1, &orbit0, &orbit1,
                           &val_time0, &val_time1, &orbit_id,
                           xo_ierr);

if (status != XO_OK)
{
  func_id = XO_ORBIT_INIT_FILE_ID;
  xo_get_msg(&func_id, xo_ierr, &n, msg);
  xo_print_msg(&n, msg);
  xl_time_close(&time_id, xl_ierr);
  if (status <= XL_ERR) return(XL_ERR);
}

```

Orbit Initialisation

```

/* Satellite Nominal Attitude frame initialisation */

/* Yaw Steering Mode */
model_enum = XP_MODEL_GENERIC;
model_param[0] = XP_NEG_Z_AXIS;
model_param[1] = XP_NADIR_VEC;
model_param[2] = 0.;
model_param[3] = 0.;
model_param[4] = 0.;
model_param[5] = XP_X_AXIS;
model_param[6] = XP_EF_VEL_VEC;
model_param[7] = 0.;
model_param[8] = 0.;
model_param[9] = 0.;
status = xp_sat_nominal_att_init_model(&model_enum, model_param,
                                       /* output */
                                       &sat_nom_trans_id, xp_ierr);

if (status != XP_OK)
{
  func_id = XP_SAT_NOMINAL_ATT_INIT_MODEL_ID;
  xp_get_msg(&func_id, xp_ierr, &n, msg);
  xp_print_msg(&n, msg);
  xo_propag_close(&propag_id, xo_ierr);
  xo_orbit_close(&orbit_id, xo_ierr);
  xl_time_close(&time_id, xl_ierr);
  if (status <= XO_ERR) return(XL_ERR);
}

```

Satellite Nominal Attitude Initialisation

```

/* Satellite Attitude frame initialisation */

ang[0] = 0.0;
ang[1] = 0.0;
ang[2] = 0.0;
status = xp_sat_att_angle_init(ang,
                                /* output */
                                &sat_trans_id,
                                xp_ierr);

if (status != XP_OK)
{
    func_id = XP_SAT_ATT_ANGLE_INIT_ID;
    xp_get_msg(&func_id, xp_ierr, &n, msg);
    xp_print_msg(&n, msg);
    xp_sat_nominal_att_close(&sat_nom_trans_id, xp_ierr);
    xo_propag_close(&propag_id, xo_ierr);
    xo_orbit_close(&orbit_id, xo_ierr);
    xl_time_close(&time_id, xl_ierr);
    if (status <= XO_ERR) return(XL_ERR);
}
    
```

Satellite Attitude Initialisation

```

/* Instrument attitude frame initialisation */

ang[0] = 0.0;
ang[1] = 0.0; /* scan angle */
ang[2] = 0.0;

offset[0] = 0.0;
offset[1] = 0.0;
offset[2] = 0.0;

status = xp_instr_att_angle_init(ang, offset,
                                  /* output */
                                  &instr_trans_id,
                                  xp_ierr);

if (status != XP_OK)
{
    [...]
}
    
```

Instrument Attitude Initialisation

```

/* Attitude initialisation */
status = xp_attitude_init (&attitude_id, xp_ierr);
if (status != XP_OK)
{
    [...]
}
    
```

Attitude Initialisation

```

/* DEM initialisation */
dem_mode = XD_DEM_ACE_MODEL;
status = xp_dem_init(&dem_mode, &dem_model, dem_file,
                    &dem_id, xp_ierr);

if (status != XP_OK)
{[...] }
    
```

DEM Initialisation

```
/* propagate along one orbit */  
  
user_time_start = 2831.00690124781;  
user_time_stop = 2831.07112143130;  
  
time_step = 500/86400.0;
```

```
for (i_loop = user_time_start; i_loop < user_time_stop; i_loop += time_step)  
{  
    time = i_loop;  
  
    /* Get satellite state vector at "time" */  
    status = xo_osv_compute(&orbit_id, &propag_model, &time_ref_utc, &time,  
                            pos, vel, acc, xo_ierr);  
    if (status != XP_OK)  
    {  
        [...]  
    }  
}
```

← Loop to get targets for different times between user_time_start and user_time_stop

Compute State Vector

```
/* Compute Attitude using the calculated state vector */  
  
target_frame = XP_INSTR_ATT;  
  
status = xp_attitude_compute(&model_id, &time_id,  
                              &sat_nom_trans_id,  
                              &sat_trans_id,  
                              &instr_trans_id,  
                              &attitude_id,  
                              &time_ref, &time,  
                              pos, vel, acc,  
                              &target_frame,  
                              xp_ierr);  
  
if (status != XP_OK)  
{  
    [...]  
}
```

Compute Attitude

```
/* Get the intersection target */

sat_id      = XP_SAT_ADM;
inter_flag  = XP_INTER_1ST;
deriv       = XL_DER_1ST;
los_az      = 90.0;
los_el      = 90.0;
los_az_rate = 1.0;
los_el_rate = 1.0;
iray        = XP_NO_REF;
freq        = 8.4e14; /* 355 nm, SPEED_OF_LIGHT = 299792458.0; [m/s] */
geod_alt    = 0.0;
num_target  = 0;

status = xp_target_inter(&sat_id,
                        &attitude_id,
                        &atmos_id,
                        &dem_id,
                        &deriv, &inter_flag, &los_az, &los_el,
                        &geod_alt,
                        &los_az_rate, &los_el_rate,
                        &iray, &freq,
                        /* output */
                        &num_user_target, &num_los_target,
                        &target_id,
                        xp_ierr);

if (status != XP_OK)
{
    [...]
}
```

Compute the target


```

/* Get User, LOS and DEM Targets Data */

for (target_type = XP_USER_TARGET_TYPE;
     target_type <= XP_DEM_TARGET_TYPE;
     target_type++)
{
    if (target_type == XP_USER_TARGET_TYPE)
        strcpy(target_name, "User target");
    if (target_type == XP_LOS_TARGET_TYPE)
        strcpy(target_name, "LOS target");
    else if (target_type == XP_DEM_TARGET_TYPE)
        strcpy(target_name, "DEM target");

    printf("\n-----\n");
    printf(" Target results for xp_target_inter and target %d\n",
target_number);
    printf(" Target type: %s. Time = %f\n", target_name, time);
    printf("-----\n");

    target_number = 1;
    choice = XL_DER_1ST;

    /* Get target parameters */
    status = xp_target_extra_vector(&target_id,
                                   &choice, &target_type, &target_number,
                                   /* output */
                                   vector_results,
                                   vector_results_rate,
                                   vector_results_rate_rate,
                                   xp_ierr);

    if (status == XP_ERR)
    {
        [...]
    }
    else
    {
        printf("\n Target extra results \n");
        printf("- Num Target      : %ld\n", targ_num);
        printf("- Target Position  : [%12.3lf,%12.3lf,%12.3lf]\n",
vector_results[0], vector_results[1], vector_results[2]);
        printf("- Target Velocity  : [%12.3lf,%12.3lf,%12.3lf]\n",
vector_results_rate[0], vector_results_rate[1],
vector_results_rate[2]);
        printf("- Range            : %lf\n",vector_results[6]);
        [...]
    }
}
    
```

← Loop to get data for the different targets

Getting target data

```

/* Get target extra main parameters */

choice = XP_TARG_EXTRA_AUX_ALL;
status = xp_target_extra_main(&target_id,
                             &choice, &target_type, &target_number,
                             main_results, main_results_rate,
                             main_results_rate_rate,
                             xp_ierr);

if (status == XP_ERR)
{
  [...]
}
else
{
  printf("\n Target extra results \n");
  printf("- Num Target           : %ld\n", targ_num);
  printf("- Geocentric Long.         : %lf\n",main_results[0]);
  printf("- Geocentric Lat.          : %lf\n",main_results[1]);
  printf("- Geodetic Latitude        : %lf\n",main_results[2]);
  [...]
}
/* Get target extra results */
choice      = XP_TARG_EXTRA_AUX_ALL;
target_number = 0;
status = xp_target_extra_aux(&target_id,
                             &choice, &target_type, &target_number,
                             aux_results, aux_results_rate, aux_results_rate_rate,
                             xp_ierr);

if (status == XP_ERR)
{
  [...]
}
else
{
  printf("\n Auxiliary Target outputs:\n");
  printf("- Curvature Radius at target's nadir = %lf\n", aux_results[0]);
  printf("- Distance: target's nadir to satellites's nadir = %lf\n",
        aux_results[1]);
  printf("- Distance target's nadir to ground track = %lf\n",
        aux_results[2]);
  printf("- Distance SSP to point in the ground track nearest to the target's
nadir= %lf\n", aux_results[3]);
  printf("- MLST at target = %lf\n", aux_results[4]);
  printf("- TLST at target = %lf\n", aux_results[5]);
  printf("- RA through the atmosphere = %lf\n", aux_results[6]);
  [...]
}

```

Getting target data

```

/* Get target-to-sun parameters */
choice      = XL_DER_1ST;
target_number = 0;
iray        = XP_NO_REF;
freq        = 1.e10;

status = xp_target_extra_target_to_sun(&target_id,
                                       &choice, &target_type, &target_number,
                                       &iray, &freq,
                                       sun_results, sun_results_rate,
                                       sun_results_rate_rate, xp_ierr);

if (status == XP_ERR)
{
    [...]
}
else
{
    printf("\n Target to Sun outputs:\n");
    printf("- Topocentric Azimuth.      : %lf\n",sun_results[0]);
    printf("- Topocentric Elevation.      : %lf\n",sun_results[1]);
    printf("- Topocentric Azimuth rate.    : %lf\n",sun_results_rate[0]);
    printf("- Topocentric Elevation rate : %lf\n",sun_results_rate[1]);
    printf("- Tangent Altitude             : %lf\n",sun_results[2]);
    printf("- Target to sun visibility.    : %g\n",sun_results[3]);
}
/* Get target-to-moon parameters */
choice      = XL_DER_1ST;
target_number = 0;
iray        = XP_NO_REF;

/* Get EF target parameters */
choice      = XL_DER_1ST;
target_number = 0;
freq        = 1.e10;
status = xp_target_extra_ef_target(&target_id,
                                   &choice, &target_type, &target_number, &freq,
                                   ef_target_results_rate,
                                   ef_target_results_rate_rate,
                                   xp_ierr);

if (status == XP_ERR)
{
    [...]
}
else
{
    printf("\n EF Target outputs:\n");
    printf("- EF target to satellite range rate : %lf\n",
          ef_target_results_rate[1]);
    printf("- EF target to satellite azimuth rate (TOP) : %lf\n",
          ef_target_results_rate[2]);
    printf("- EF target to satellite elevation rate (TOP) : %lf\n",
          ef_target_results_rate[3]);
    [...]
}
} /* end for "target_type" (End loop to get data for the different targets)*/
    
```

Getting target data

```
/* Closing Ids */  
    status = xp_target_close(&target_id, xp_ierr);  
    [...]
```

Closing
target

```
} /* end for "i_loop" (End loop for different times)*/
```

```
status = xp_attitude_close(&attitude_id, xp_ierr);  
[...]
```

Close
attitude

```
status = xp_sat_nominal_att_close(&sat_nom_trans_id, xp_ierr);  
[...]
```

```
status = xp_sat_att_close(&sat_trans_id, xp_ierr);  
[...]
```

```
status = xp_instr_att_close(&instr_trans_id, xp_ierr);  
[...]
```

Close satellite
attitude frame

```
status = xp_dem_close(&dem_id, xp_ierr);  
[...]
```

Close
DEM

```
status = xo_orbit_close(&orbit_id, xo_ierr);  
[...]
```

Close
orbit

```
status = xl_time_close(&time_id, xl_ierr);  
[...]
```

Close
time

```
}
```

```
/* end */
```

4.18 Swath calculations

A swath can be defined as the track swept by the field of view of an instrument in the satellite along a time interval. For the aim of this section this definition is enough, however the definition of a swath can be much more complex. For a detailed definition about swaths refer to [VIS_SUM] section 7.1.2.

The Earth Observation CFI software can handle swath data using two different data sets provided by :

- Swath Definition files (SDF): they contain information about the swath type and geometry and the satellite attitude.

or

- Swath Template files (STF): they contain the list of longitude and latitude points of the swath if the orbit started at longitude and latitude 0.

The format for the two files can be found in [D_H_SUM].

Swath files are mainly useful for the the visibility calculations (section 4.19) but the CFI software provides other functions for getting information from swaths:

- Reading and writing swath files (see section 4.3, 4.4 and [D_H_SUM]): These functions allow the user to read a swath file and store the information in a data structure (reading functions) or to dump to a file the swath data contained in a structure (writing function).
- Generate a STF from a SDF (function `xv_gen_swath` described in [VIS_SUM]): this operation requires the initialisation of the `xo_orbit_id` (section 4.13) and optionally the `atmos_id` (4.17.2) if the swath has to take into account the ray path refraction by the atmosphere.
- Calculating the swath position for a given time (function `xv_swath_pos` described in [VIS_SUM]): This operation requires the initialisation of the `xo_orbit_id` and the data structure containing the swath points from a STF (read with `xd_read_stf`).

The following figure shows an schema for the calling sequence for the described operations:

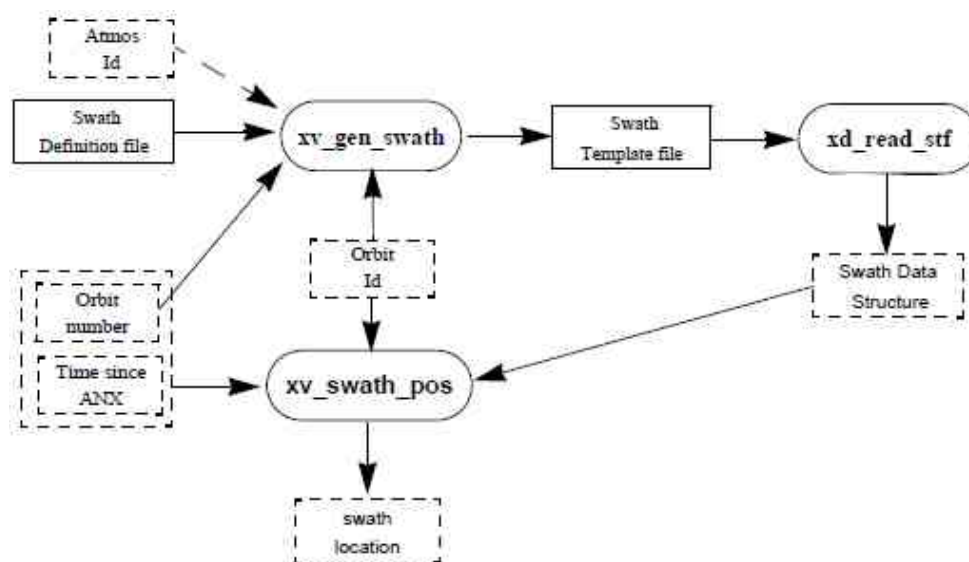


Figure 11: EO_VISIBILITY Data Flow- Swath

Note that in order to produce consistent data the same `xo_orbit_id` is used in the two calls of the swath functions.

Also the orbit number introduced in `xv_gen_swath` is the same orbit number that is passed to `xv_swath_pos`. This is not mandatory but advisable. `xv_gen_swath` produce the STF taken into account the orbit geometry so it produces the same file for all orbits with the same geometry (for example, all the orbits within the same orbital change in an OSF). In consequence, there is not need of generating a new STF every time that `xv_swath_pos` is going to be called for a different orbit, only it is needed if the orbit geometry changes.

Example 4.17 - I: Getting the swath position

```

/* Variables */
long   status, func_id, n;
char   msg[XL_MAX_COD][XL_MAX_STR];
long   xl_ierr[XL_ERR_VECTOR_MAX_LENGTH];
long   xd_ierr[XD_ERR_VECTOR_MAX_LENGTH];
long   xo_ierr[XO_ERR_VECTOR_MAX_LENGTH];
long   xv_ierr[XV_ERR_VECTOR_MAX_LENGTH];

long   sat_id      = XO_SAT_CRYOSAT;
xl_time_id time_id = {NULL};
xl_model_id model_id = {NULL};
xo_orbit_id orbit_id = {NULL};
xp_atmos_id atmos_id = {NULL};

double tri_time[4],
       tri_orbit_num = 10, /* dummy */
       tri_anx_time  = 5245.123456, /* dummy */
       tri_orbit_duration = 6035.928144; /* dummy */

long n_files, time_mode, orbit_mode, time_ref;
char orbit_scenario_file[XD_MAX_STR];
char *files[2];

long req_orbit;
char dir_name[256];
char sdf_name[256], stf_name[256];
char file_class[] = "TEST";
long version_number = 1;
char fh_system[] = "CFI";

xd_stf_file stf_data;
long orbit_type, abs_orbit, second, microsec, cycle;
double long_swath, lat_swath, alt_swath;

```

Variable declaration

```

/* Time initialisation */
tri_time[0] = -245.100000000; /* TAI time [days] */
tri_time[1] = tri_time[0] - 35.0/86400.; /* UTC time [days] (= TAI - 35.0 s) */
tri_time[2] = tri_time[0] - 35.3/86400.; /* UT1 time [days] (= TAI - 35.3 s) */
tri_time[3] = tri_time[0] - 19.0/86400.; /* GPS time [days] (= TAI - 19.0 s) */

status = xl_time_ref_init(tri_time, &tri_orbit_num, &tri_anx_time,
                        &tri_orbit_duration, &time_id, xl_ierr);
[ ...error handling for xl_time_ref_init... ]

```

Time Initialisation

```
/* Orbit initialisation: xo_orbit_init_file */
n_files = 1;
time_mode = XO_SEL_FILE;
orbit_mode = XO_ORBIT_INIT_OSF_MODE;
time_ref = XO_TIME_UT1;
strcpy(orbit_scenario_file, "./CRYOSAT_XML_OSF");
files[0] = orbit_scenario_file;

status = xo_orbit_init_file(&sat_id, &model_id, &time_id,
                           &orbit_mode, &n_files, files,
                           &time_mode, &time_ref,
                           &time0, &time1, &orbit0, &orbit1,
                           &val_time0, &val_time1,
                           &orbit_id, xo_ierr);

[ ...error handling for xo_orbit_init_file...]
```

Orbit Initialisation

```
/* Generate Swath Template file */

req_orbit = 150;
strcpy(sdf_name, "./SDF_MERIS.EEF"); /* SDF */
strcpy(dir_name, ""); /* -> generate file in current directory */
strcpy(stf_name, "EXAMPLE_STF.EEF");

status = xv_gen_swath(&orbit_id, &atmos_id, &req_orbit,
                     sdf_name, dir_name, stf_name,
                     file_class, &version_number, fh_system,
                     xv_ierr);

if (status != XV_OK)
{
    func_id = XV_GEN_SWATH_ID;
    xv_get_msg(&func_id, xv_ierr, &n, msg);
    xv_print_msg(&n, msg);
}
```

Generate Swath Template file

```
/* Read Swath Template file */
status = xd_read_stf(stf_name, &stf_data, xd_ierr);
if (status != XV_OK)
{
    func_id = XD_READ_STF_ID;
    xd_get_msg(&func_id, xd_ierr, &n, msg);
    xv_print_msg(&n, msg);
}
```

Read STF

```

orbit_type = XV_ORBIT_ABS;
abs_orbit = 2950;
second = 100;
microsec = 500000;

status=xv_swath_pos(&orbit_id, &stf_data,
                  &orbit_type, &abs_orbit, &second, &microsec, &cycle,
                  &long_swath, &lat_swath, &alt_swath,
                  xv_ierr);

if (status != XV_OK)
{
    func_id = XV_SWATH_POS_ID;
    xv_get_msg(&func_id, xv_ierr, &n, msg);
    xv_print_msg(&n, msg);
}

/* print outputs */
printf("Input absolute Orbit + time since ANX: %d + %lf s\n",
       abs_orbit, second+(microsec*1.e-6));
printf("Outputs: \n");
printf("Swath point (longitude, latitude, altitude): (%lf, %lf, %lf) \n",
       long_swath, lat_swath, alt_swath);
    
```

Get swath position

```

/* free memory for the STF */
xv_free_stf(&stf_data);
    
```

Free STF

```

/* Close orbit_id */
status = xo_orbit_close(&orbit_id, xo_ierr);
if (status != XO_OK)
{
    func_id = XO_ORBIT_CLOSE_ID;
    xo_get_msg(&func_id, xo_ierr, &n, msg);
    xo_print_msg(&n, msg);
}
    
```

Close orbit

```

/* close time reference */
status = xl_time_close(&time_id, xl_ierr);
if (status != XO_OK)
{
    func_id = XL_TIME_CLOSE_ID;
    xo_get_msg(&func_id, xl_ierr, &n, msg);
    xo_print_msg(&n, msg);
}
    
```

Close time

[...]

4.19 Visibility calculations

The Earth Observation CFI software contains a set of functions to compute the time intervals in which a satellite instrument has visibility of :

- an Earth zone
- a ground station
- a data relay satellite (DRS)
- a star

Visibility segments are provided as an orbit number plus the time since the ANX.

In order to calculate the visibility time intervals the functions require as inputs:

- orbital information provided via an orbit Id (see section 4.13)
- requested orbit interval in which the visibilities are to be computed.
- Swath information except for the DRS visibility: It can be a swath definition file or a swath template file.
- Information about the target: zone, station, DRS or the star.

The following figure shows a possible calling sequence for visibility calculation:

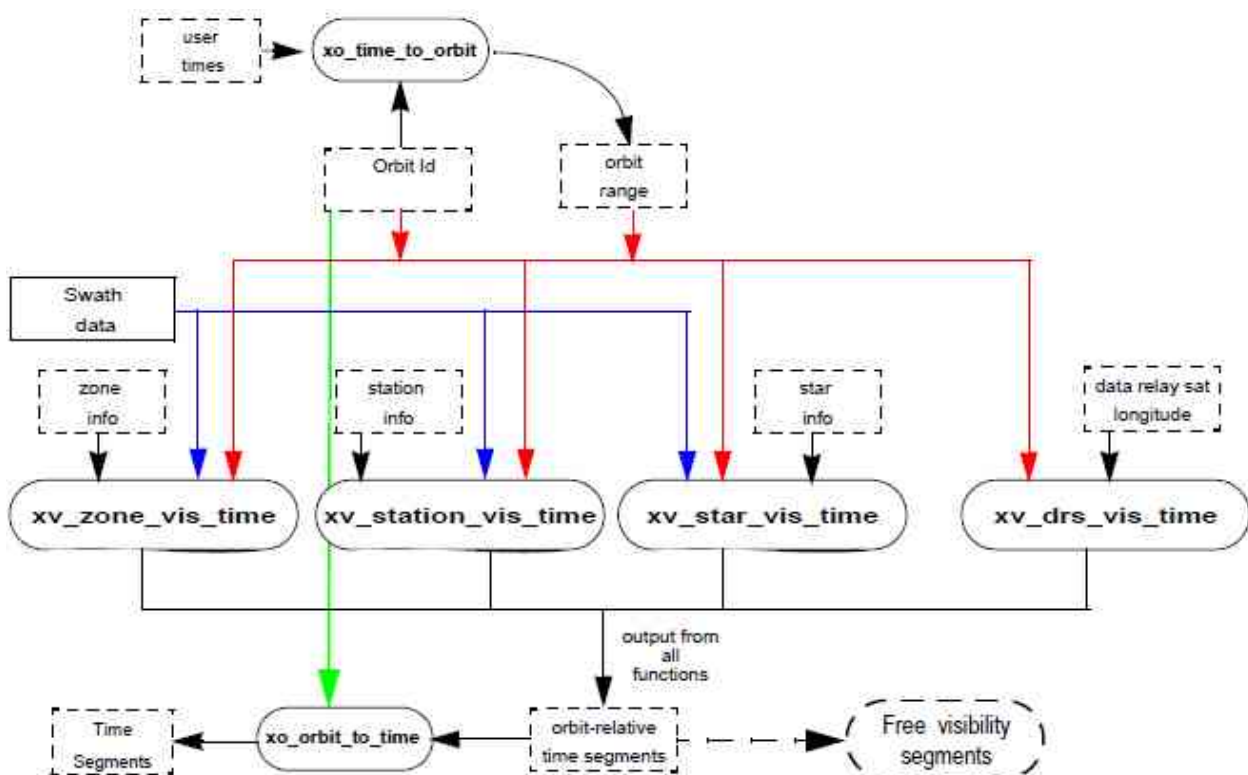


Figure 12: EO_VISIBILITY Data Flow - Visibility

Details about the visibility functions can be found in [VIS_SUM].

For those functions that require swath data, note that it can be provided by a SDF or a STF. The file type has to be indicated with an input flag (swath_flag):

- if swath_flag is zero, then the input file is a STF. Visibility segments will be computed with that file for all the requested orbits.

- if swath flag is greater than zero, then the input file is a SDF. The function will compute automatically the swath points. There are two possibilities:
 - The input `xo_orbit_id` was generated with an orbit scenario file or with `xo_orbit_init_def`: the swath points are generated only once for the first requested orbit. The visibility segments are computed with those swath points for all the orbits.
 - The input `xo_orbit_id` was generated with orbit state vectors: the swath points are generated for every `n` orbits, where `n` is the value of the `swath_flag` variable.

All the visibility functions return the segments as dynamical arrays, so when they are not to be used any more, the arrays should be freed.

Example 4.18 - I: Getting visibility segments for a zone

```

/* Variables */
long  status, func_id, n;
char  msg[XL_MAX_COD][XL_MAX_STR];
long  xl_ierr[XL_ERR_VECTOR_MAX_LENGTH];
long  xo_ierr[XO_ERR_VECTOR_MAX_LENGTH];
long  xv_ierr[XV_ERR_VECTOR_MAX_LENGTH];

long  sat_id      = XO_SAT_CRYOSAT;
xl_time_id time_id = {NULL};
xl_model_id model_id = {NULL};
xo_orbit_id orbit_id = {NULL};

[... variables for time and orbit initialisation...]

long orbit_type, start_orbit, stop_orbit,
    start_cycle, stop_cycle;
long swath_flag;
char swath_file[256];
char zone_id[9], zone_db_file[XV_MAX_STR];
long projection, zone_num;
double zone_long[10], zone_lat[10], zone_diam;
long number_segments;
long *bgn_orbit, *bgn_second, *bgn_microsec, *bgn_cycle,
    *end_orbit, *end_second, *end_microsec, *end_cycle,
    *coverage;
    
```

Variable declaration

```

/* Time initialisation */
tri_time[0] = -245.100000000; /* TAI time [days] */
tri_time[1] = tri_time[0] - 35.0/86400.; /* UTC time [days] (= TAI - 35.0 s) */
tri_time[2] = tri_time[0] - 35.3/86400.; /* UT1 time [days] (= TAI - 35.3 s) */
tri_time[3] = tri_time[0] - 19.0/86400.; /* GPS time [days] (= TAI - 19.0 s) */

status = xl_time_ref_init(tri_time, &tri_orbit_num, &tri_anx_time,
    &tri_orbit_duration, &time_id, xl_ierr);
[ ...error handling for xl_time_ref_init... ]
    
```

Time Initialisation

```
/* Orbit initialisation: xo_orbit_init_file */  
n_files = 1;  
time_mode = XO_SEL_FILE;  
orbit_mode = XO_ORBIT_INIT_OSF_MODE;  
time_ref = XO_TIME_UT1;  
strcpy(orbit_scenario_file, "./CRYOSAT_XML_OSF");  
files[0] = orbit_scenario_file;  
  
status = xo_orbit_init_file(&sat_id, &model_id, &time_id,  
                           &orbit_mode, &n_files, files,  
                           &time_mode, &time_ref,  
                           &time0, &time1, &orbit0, &orbit1,  
                           &val_time0, &val_time1,  
                           &orbit_id, xo_ierr);  
  
[ ...error handling for xo_orbit_init_file...]
```

Orbit Initialisation

```
/* Calling xv_zone_vis_time */  
orbit_type = XV_ORBIT_ABS;  
start_orbit = 2900;  
stop_orbit = 2950;  
  
strcpy(swath_file, "./RA_2_SDF_N1"); /* SDF */  
strcpy(zone_id, "ZANA_____");  
strcpy(zone_db_file, "./ZONE_FILE.EEF");  
  
projection = 0;  
zone_num = 0; /* To be able to introduce the zone identifications */  
min_duration = 0.0;  
  
status = xv_zone_vis_time(&orbit_id, &orbit_type,  
                          &start_orbit, &start_cycle,  
                          &stop_orbit, &stop_cycle,  
                          &swath_flag, swath_file,  
                          zone_id, zone_db_file,  
                          &projection, &zone_num,  
                          zone_long, zone_lat, &zone_diam,  
                          &min_duration,  
                          &number_segments,  
                          &bgn_orbit, &bgn_second,  
                          &bgn_microsec, &bgn_cycle,  
                          &end_orbit, &end_second,  
                          &end_microsec, &end_cycle,  
                          &coverage, xv_ierr);  
  
if (status != XV_OK)  
{  
    func_id = XV_ZONE_VIS_TIME_ID;  
    xv_get_msg(&func_id, xv_ierr, &n, msg);  
    xv_print_msg(&n, msg);  
}
```

Getting visibility segments

```

/* print outputs */
printf("Inputs: \n");
printf("  Start/Stop Absolute Orbit: %d / %d\n", start_orbit, stop_orbit);
printf("  Zone File: %s\n", zone_db_file);
printf("  Zone Id: %s\n", zone_id);
printf("Outputs: \n");
printf("Number of segments: %d\n", number_segments);
printf("  Segments: Start (Orbit, seconds, microseconds) --
        Stop (Orbit, seconds, microseconds)\n");

for(i=0; i < number_segments; i++)
{
  printf("          (%4d, %4d, %6d) -- (%4d, %4d, %6d)\n",
        bgn_orbit[i], bgn_second[i], bgn_microsec[i],
        end_orbit[i], end_second[i], end_microsec[i]);
}

```

Print visibility segments

```

/* free memory: The cycle are not allocated as the orbit type
is absolute orbits*/
free(bgn_orbit);
free(bgn_secs);
free(bgn_microsecs);

free(end_orbit);
free(end_second);
free(end_microsec);
free(coverage);

```

Free memory for the visibility segments

```

/* Close orbit_id */
status = xo_orbit_close(&orbit_id, xo_ierr);
if (status != XO_OK)
{
  func_id = XO_ORBIT_CLOSE_ID;
  xo_get_msg(&func_id, xo_ierr, &n, msg);
  xo_print_msg(&n, msg);
}

```

Close orbit

```

/* close time reference */
status = xl_time_close(&time_id, xl_ierr);
if (status != XO_OK)
{
  func_id = XL_TIME_CLOSE_ID;
  xo_get_msg(&func_id, xl_ierr, &n, msg);
  xo_print_msg(&n, msg);
}

```

Close time

[...]

4.20 Time segments manipulation

The EO_VISIBILITY library provides a set of functions for doing logical operations between sets of time segments. A time segment is given by an absolute or relative orbit number plus the time since the ANX for the entry and the exit of the segment, this way the functions can handle the segments coming from the output of the visibility functions.

These operations are:

- Getting the complement of a list of time segments (**xv_time_segments_not**).
- Getting the intersection of two lists of time segments (**xv_time_segments_and**).
- Getting the union of two lists of time segments (**xv_time_segments_or**).
- Adding or subtracting time durations at the beginning and end of every time segment within a list (**xv_time_segments_delta**).
- Sorting a list of time segments (**xv_time_segments_sort**).
- Merging all the overlapped segments in a list (**xv_time_segments_merge**).
- Getting a subset of the time segments list, such that this subset covers entirely a zone or line swath (**xv_time_segments_mapping**).

A detailed explanation of these functions is in [VIS_SUM].

In order to use the functions, the following strategy has to be followed:

- The orbit initialisation is required if the input segments are given in relative orbits. Normally, if the time segments come from visibility functions, the *xo_orbit_id* structure will be already initialised.
- Call the required function for segment manipulation.
- The output time segments are returned as dynamical arrays, so when they are not going to be used any more, the arrays should be freed.

Example 4.19 - I: Time segments manipulation (Intersection example)

```
/* Variables */
long orbit_type, start_orbit, stop_orbit,
    start_cycle, stop_cycle;
long swath_flag;
char swath_file[256];
char zone_id[9], zone_db_file[XV_MAX_STR];
long projection, zone_num;
double zone_long[10], zone_lat[10], zone_diam;

long num_1;
long *bgn_orbit_1, *bgn_secs_1, *bgn_microsecs_1, *bgn_cycle_1,
    *end_orbit_1, *end_secs_1, *end_microsecs_1, *end_cycle_1;
long coverage_1;

long num_2;
long *bgn_orbit_2, *bgn_secs_2, *bgn_microsecs_2, *bgn_cycle_2,
    *end_orbit_2, *end_secs_2, *end_microsecs_2, *end_cycle_2;
long coverage_2;

long num_out, order_switch;
long *bgn_orbit_res, *bgn_secs_res, *bgn_microsecs_res, *bgn_cycle_res,
    *end_orbit_res, *end_secs_res, *end_microsecs_res, *end_cycle_res;
long order_criteria;
long xv_ierr[XV_ERR_VECTOR_MAX_LENGTH];
```

```
/* Time and orbit initialization */
[...]
```

Variable declaration

Time and orbit
initialisation

```
/* Getting visibility segments for zone 1 */
orbit_type = XV_ORBIT_ABS;
start_orbit = 2900;
stop_orbit = 2950;
strcpy(swath_file, "./RA_2_SDF.N1"); /* SDF */
strcpy(zone_id, "ZONE_1");
strcpy(zone_db_file, "./ZONE_FILE.EEF");

projection = 0;
zone_num = 0; /* To be able to introduce the zone identifications */
min_duration = 0.0;

status = xv_zone_vis_time(&orbit_id, &orbit_type,
                        &start_orbit, &start_cycle,
                        &stop_orbit, &stop_cycle,
                        &swath_flag, swath_file,
                        zone_id, zone_db_file,
                        &projection, &zone_num,
                        zone_long, zone_lat, &zone_diam,
                        &min_duration,
                        &num_1,
                        &bgn_orbit_1, &bgn_secs_1,
                        &bgn_microsecs_1, &bgn_cycle_1,
                        &end_orbit_1, &end_secs_1,
                        &end_microsecs_1, &end_cycle_1,
                        &coverage, xv_ierr);

[... Error handling...]

/* Getting visibility segments for zone 2*/
strcpy(zone_id, "ZONE_2");

status = xv_zone_vis_time(&orbit_id, &orbit_type,
                        &start_orbit, &start_cycle,
                        &stop_orbit, &stop_cycle,
                        &swath_flag, swath_file,
                        zone_id, zone_db_file,
                        &projection, &zone_num,
                        zone_long, zone_lat, &zone_diam,
                        &min_duration,
                        &num_2,
                        &bgn_orbit_2, &bgn_secs_2,
                        &bgn_microsecs_2, &bgn_cycle_2,
                        &end_orbit_2, &end_secs_2,
                        &end_microsecs_2, &end_cycle_2,
                        &coverage, xv_ierr);

[... Error handling...]
```

Getting visibility segments for two different zones.
Afterwards the intersection between the two sets of segments will be computed

```
/* Getting the intersection */
order_switch = XV_TIME_ORDER; /* flag to indicate that the input segments are
                                already ordered. It saves computation time */

status=xv_time_segments_and (&orbit_id,
                             &orbit_type, &order_switch,
                             /* input segments list 1*/
                             &num_1,
                             bgn_orbit_1, bgn_secs_1,
                             bgn_microsecs_1, bgn_cycle_1,
                             end_orbit_1, end_secs_1,
                             end_microsecs_1, end_cycle_1,
                             /* input segments list 2*/
                             num_2,
                             bgn_orbit_2, bgn_secs_2,
                             bgn_microsecs_2, bgn_cycle_2,
                             end_orbit_2, end_secs_2,
                             end_microsecs_2, end_cycle_2,
                             /* output segments list */
                             &num_out,
                             &bgn_orbit_res, &bgn_secs_res,
                             &bgn_microsecs_res, &bgn_cycle_res,
                             &end_orbit_res, &end_secs_res,
                             &end_microsecs_res, &end_cycle_res,
                             xv_ierr);

if (status != XV_OK)
{
    func_id = XV_TIME_SEGMENTS_AND_ID;
    xv_get_msg(&func_id, xv_ierr, &n, msg);
    xv_print_msg(&n, msg);
}

/* print outputs */
printf("Outputs for segment intersection: \n");
printf("  Number of segments: %d\n", num_out);
printf("  Segments: Start (Orbit, seconds, microseconds) --
        Stop (Orbit, seconds, microseconds)\n");

for(i=0; i < num_out; i++)
{
    printf("          (%4d, %4d, %6d) -- (%4d, %4d, %6d)\n",
           bgn_orbit_res[i], bgn_secs_res[i], bgn_microsecs_res[i],
           end_orbit_res[i], end_secs_res[i], end_microsecs_res[i]);
}
}
```

Getting segment intersection for the two sets of segments


```
/* Freeing the memory */  
free(bgn_orbit_res);  
free(bgn_secs_res);  
free(bgn_microsecs_res);  
free(bgn_cycle_res);  
  
free(end_orbit_res);  
free(end_secs_res);  
free(end_microsecs_res);  
free(end_cycle_res);  
  
free(bgn_orbit_1);  
free(bgn_secs_1);  
free(bgn_microsecs_1);  
  
free(end_orbit_1);  
free(end_secs_1);  
free(end_microsecs_1);  
  
free(bgn_orbit_2);  
free(bgn_secs_2);  
free(bgn_microsecs_2);  
  
free(end_orbit_2);  
free(end_secs_2);  
free(end_microsecs_2);
```

Free memory

```
/* Closing orbit and time Ids. */  
[...]
```

Close
Ids

```
[...]
```